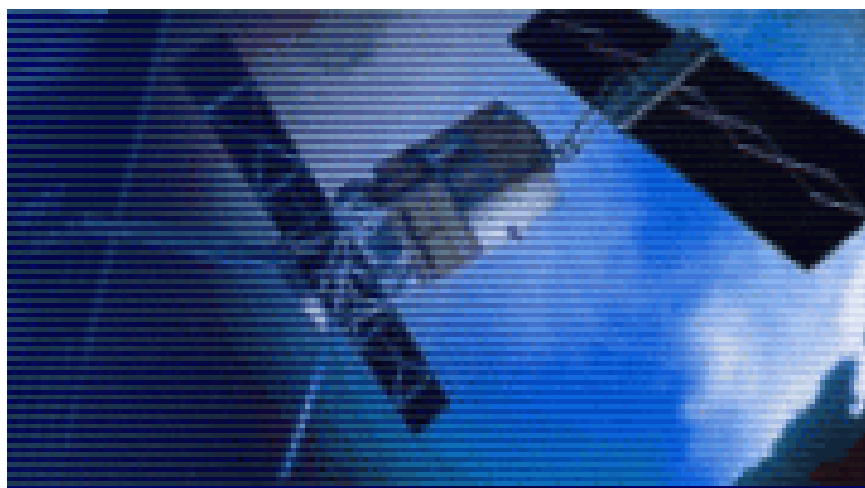


# ERS-2 Wind Scatterometer Cyclic Report

From 2<sup>nd</sup> May to 24<sup>th</sup> October 2005  
Cycles 105 – 106- 107 – 108 - 109



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Prepared by:	A. Paciucci	(scat@dpqc.org)
Approved by:	R. Crapolichio	(scat@dpqc.org)
Checked by:	S. Procter	(qc@dpqc.org)
Inputs from:	H. Hersbach	(ECMWF)

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## **1 Introduction and Summary**

The document includes a summary of the daily quality control made within the PCS and various sections describing the results of the investigations and studies of “open-problems” related to the Scatterometer. In each section results are shown from the beginning of the mission in order to see the evolution and to outline possible “seasonal” effects. An explanation for the major events which have impacted the performance since launch is given, and comments about the recent events which occurred during the last cycle are included. This report covers the period from 2<sup>nd</sup> May 2005 to 24<sup>th</sup> October 2005 (cycles 105-106-107-108-109) and includes the results of the monitoring activity performed by ESRIN.

- This document is available on line: [http://earth.esa.int/pcs/ers/scatt/reports/pcs\\_cyclic/](http://earth.esa.int/pcs/ers/scatt/reports/pcs_cyclic/)

### **Mission events**

- The ERS-2 satellite was piloted in ZGM throughout in all cycles 105 to 109.
- During cycles 105 - 109 the ESACA processor worked nominally without faults.
- AMI instrument was switched off due to Payload switched off after RA anomaly between 23<sup>rd</sup> May 2005 18:31:02 and 24<sup>th</sup> May 2005 15:06:40 for cycle 105; during cycle 106 the AMI was switched down caused by RBI status error from 20<sup>th</sup> June at 20:44:26 to 21<sup>st</sup> June 2005 at 10:13:28; during the cycle 107 we had the following AMI unavailability due to RA synchronization on 01<sup>st</sup> Aug 2005 from 13:37:54 to 13:52:08, from 15:06:51 to 15:12:10 and from 16:46:09 to 17:00:44; during the cycle 108 AMI instrument was in standby mode on 15<sup>th</sup> September 2005 from 15:05:14 to 20:04:25; and for cycle 109 AMI instrument was in standby mode on 28<sup>th</sup> September 2005 between 15:21:22 and 21:33:13.
- Fast delivery UWI data produced during the following periods has been generated without meteorological background information: between 15<sup>th</sup> – 18<sup>th</sup> August 2005 due to ESRIN network problem; for the period 22<sup>nd</sup> – 23<sup>rd</sup> August 2005 and 28<sup>th</sup> – 29<sup>th</sup> August 2005 due to PCS System problem. This caused a degradation of the wind field quality with a temporary de-aliasing problem of UWI products
- For the entire period in cycles 105 - 109, ERS-2 Scatterometer data was used in the 4D-Var data assimilation system at ECMWF.
- News on ERS mission is available on line: [http://earth.esa.int/ers/new\\_ers\\_news.html](http://earth.esa.int/ers/new_ers_news.html)

### **Yaw performance**

- The result of the yaw monitoring for cycles 105, 106, 107, 108, 109 is a yaw error angle within the expected nominal range (+/- 2 degrees) with an average level around 0 deg. for most of the orbits. An increase of degraded yaw performance was noted starting during cycle 107.

### **Calibration performance**

- Calibration data from Transponder are regularly acquired and archived for re-processing. Calibration results will be provided in the next reports.
- Due to the regional mission scenario the calibration performances over the Brazilian rain forest are not available because that area is not covered by the ESA ground station. The chance to install a new station to cover the calibration site is under investigation as well as

the possibility to use stable ice area in Greenland and Antarctica to monitor the instrument calibration.

- The Ocean Calibration monitoring is performed by ECMWF and all the information related to this is available on web (<http://earth.esa.int/pcs/ers/scatt/reports/ecmwf> )

### **Instrument performance**

- During the cycle 105 – 109 the behavior of the mean transmitted power evolution had a decrease of 0.08 db/cycle. That value is closed to the nominal trend detected since August 1996 (about 0.1 dB per cycle). Only for cycle 107 we note an average increase do not confirmed in the following cycles.
- The evolution of the noise power during the cycle 105 - 109 was stable (see Figure 3). The daily average for the Fore and Aft beam noise is around 1.7 ADC (I) and around 1.6 ADC (Q) respectively. For the Mid beam the noise is not measurable.
- During the cycles 105 - 109 the Doppler compensation evolution was very stable. The CoG of the compensated signal was around 0 Hz for the Fore and Aft antenna and around 200 Hz for the Mid antenna. The standard deviation of the CoG was around 1500 Hz for the Fore and Aft antenna and around 2700 Hz for the Mid antenna. The small variation on the CoG evolution around 31<sup>st</sup> May is due to a slight degradation of the satellite pointing. The Data gap in the statistics between 26 - 29 August 2005 was due to a PCS System problem.

### **Product performance**

During cycle 105 -109 the data was received at ECMWF between 21:05 UTC 2 May 2005 and 20:57 UTC 24 October 2005. For any information related to data at ECMWF refer to cyclic reports published on web at <http://earth.esa.int/pcs/ers/scatt/reports/ecmwf> .

The PCS geophysical monitoring reports wind performances very stable except for the period between 15<sup>th</sup> – 18<sup>th</sup> August 2005 due to ESRIN network problem, and for the 22<sup>nd</sup> – 23<sup>rd</sup> August 2005 and 28<sup>th</sup> – 29<sup>th</sup> August 2005 due to a PCS System problem. This caused a degradation of the wind field quality with a temporary de-aliasing problem of UWI products. For the rest of the period the wind speed bias (UWI vs 18 or 24 hour forecast) was closed in nominal trend (around 0.7 m/s) and the speed bias standard deviation was in nominal seasonal trend around 2 m/s.

The wind direction deviation was good. The performance for UWI wind direction follows the seasonal trend. The standard deviation of UWI wind speed has increased, the reason could be related to a less mild wind climate. For all the information related to ECMWF refer to web page <http://earth.esa.int/pcs/ers/scatt/reports/ecmwf>

## **2 Calibration Performances**

The calibration performances are estimated using three types of target: a man-made target (the transponder) and two natural targets (the rain forest and the ocean). This approach allows us to design the correct calibration using punctual but accurate information from transponders and extended but noisy information from rainforest and ocean for which the main component of the variance comes from the geophysical evolution of the natural target and from the backscattering models used. These aspects are in the calibration performance monitoring philosophy. The major goals of the calibration monitoring activities are the achievement of a “flat” antenna pattern profile and the assurance of a stable absolute calibration level.

### **2.1 Gain Constant over transponder**

One gain constant is computed per transponder per beam from the actual and simulated two-dimensional echo power, which is given as a function of the orbit time and range time. This parameter clearly indicates the difference between the “real instrument” and the mathematical model. In order to acquire data over the transponder the Scatterometer must be set in an appropriate operational mode defined as “Calibration Mode”. Since January 2001 with the operations in Zero Gyro Mode (ZGM) the satellite attitude is not as stable as it was in the nominal Yaw Steering Mode (YSM). In particular there is a non-predictable variation of the yaw error angle along the orbit. For that reason the gain constant data computed by the CALPROC processor, that assumes a stable orbit, are meaningless and a new calibration processor is under development. In the mean time, data from the Transponder are still acquired and archived for future re-processing. The reprocessed gain constants will be provided in this section when available. For the gain constant computed during the nominal YSM please refer to the Scatterometer cyclic report cycle 60.

### **2.2 Ocean Calibration**

The average sigma0 bias levels (compared to simulated sigma0's based on ECMWF model FG winds) stratified with respect to antenna beam, ascending or descending track and as function of incidence angle (i.e. across-node number) is displayed and analyzed in ECMWF reports available on web (see <http://earth.esa.int/pcs/ers/scatt/reports/ecmwf> for more details)

### 2.3 Gamma-nought over the Brazilian rainforest

Although the transponders give accurate measurements of the antenna attenuation at particular points of the antenna pattern, they are not adequate for fine tuning across all incidence angles, as there are simply not enough samples. The tropical rainforest in South America has been used as a reference distributed target. The target at the working frequency (C-band) of ERS-2 Scatterometer acts as a very rough surface, and the transmitted signal is equally scattered in all directions (the target is assumed to follow the isotropic approximation). Consequently, for the angle of incidence used by ERS-2 Scatterometer, the normalized backscattering coefficient (sigma nought) will depend solely on the surface effectively seen by the instrument:

$$S^0 = S \cdot \cos \theta$$

With this hypothesis it is possible to define the following formula:

$$\gamma^0 = \frac{\sigma^0}{\cos \theta}$$

Using the above equation, the gamma nought backscattering coefficient over the rain forest is independent of the incident angle, allowing the measurements from each of the three beams to be compared. The test area used by the PCS is located between 2.5 degrees North and 5.0 degrees south in latitude and 60.5 degrees West and 70.0 degrees West in longitude. That area is actually not covered by the Regional mission scenario (since cycle 86 onwards) and therefore the calibration monitoring activity over the Brazilian rainforest is suspended because no data are available. The chance to continue the monitoring activity with a new receiving station covering the Brazilian rainforest is under investigation. The following paragraphs will report on the results when data will be available.

### 2.4 Antenna pattern: Gamma-nought as a function of elevation angle

Due to the regional mission scenario data over the Brazilian rainforest are not available. For that reason the antenna patterns as a function of the elevation angle have not been computed.

### 2.5 Antenna pattern: Gamma-nought as a function of incidence angle

Due to the regional mission scenario data over the Brazilian rainforest are not available. For that reason the antenna patterns as a function of the incidence angle have not been computed.

### 2.6 Gamma nought histograms and peak position evolution

As the gamma nought is independent from the incidence angle, the histogram of gamma nought over the rainforest is characterized by a sharp peak. The time-series of the peak position gives some information on the stability of the calibration. This parameter is computed by fitting the histogram with a normal distribution added to a second order polynomial:

$$F(x) = A_0 \cdot \exp\left(-\frac{z^2}{2}\right) + A_3 + A_4 \cdot x + A_5 \cdot x^2$$

where:  $z = \frac{x - A_1}{A_2}$

The parameters are computed using a non linear least square method called “gradient expansion”. The position of the peak is given by the maximum of the function F(x). The histograms are computed weekly (from Monday to Sunday) for each antenna individually “Fore”, “Mid” and “Aft” and for ascending and descending passes with a bin size of 0.02 dB. Due to the regional mission scenario data over the Brazilian rainforest are not available and the histograms have not been computed. For the time series since the beginning of the mission please refer to the Scatterometer cyclic report cycle 86.

## **2.7 Gamma nought image of the reference area**

Due to the regional mission scenario data over the Brazilian rainforest are not available and the histograms have not been computed.

## **2.8 Sigma nought evolution**

Due to the regional mission scenario data over the Brazilian rainforest are not available. For that reason no update has been made to the sigma nought evolution time series. For the time series since the beginning of the mission until June 2003 please refer to the Scatterometer cyclic report cycle 86.

## **2.9 Antenna temperature evolution over the Rain Forest**

Due to the regional mission scenario data over the Brazilian rainforest are not available. For the time series since the beginning of the mission please refer to the Scatterometer cyclic report cycle 86.

### **3 Instrument performance**

The instrument status is checked by monitoring the following parameters:

- Centre of Gravity (CoG) and standard deviation of the received signal spectrum after the on-ground Doppler Compensation filter. This parameter is useful for the monitoring of the orbit stability, the performances of the Doppler compensation filter, the behavior of the yaw steering mode and the performances of the devices in charge for the satellite attitude (e.g. gyroscopes, Earth sensor, Sun sensor).
- Noise power I and Q channel.
- Internal calibration pulse power.

The latter is an important parameter to monitor the transmitter and receiver chain, the evolution of pulse generator, the High Power Amplifier (HPA), the Traveling Wave Tube (TWT) and the receiver. These parameters are extracted daily from the UWI products and averaged. The evolution of each parameter is characterized by a least square line fit. The coefficients of the line fit are printed in each plot.

#### **3.1 Centre of gravity and standard deviation of received power spectrum**

The Figure 1 shows the evolution of the two parameters for each beam since the beginning of the ERS-2 mission.

The tendency during the nominal Yaw Steering Mode (YSM) period (beginning of the mission since the operation with the Mono Gyro (MGM) Attitude On-board Control System (AOCS) configuration on 7<sup>th</sup> February 2000) is a small and regular increase of the Centre of gravity (CoG) of received spectrum for the three antennae. During the YSM, two small changes can be detected in the CoG evolution. The first change is from 24<sup>th</sup>, January 1996 to 14<sup>th</sup>, March 1996, the second one is from 14<sup>th</sup> February 1997 to 22<sup>nd</sup> April 1997. The reason was a change in the pointing subsystem (DES reconfiguration) side B instead of side A after a depointing anomaly (see table 1 for the list of the all AOCS depointing anomaly occurred during the ERS-2 mission). During these periods side B was switched on. It is important to note that during the first time a clear difference in the CoG of the received spectrum is present only for the Fore antenna (an increase of roughly 100 Hz) while during the second time the change has affected all the three antennae (roughly an increase of 200 Hz, 50 Hz and 50 Hz for the fore, mid and aft antenna respectively).

At the beginning of 2000 the nominal 3-gyros AOCS configuration (plus one Digital Earth Sensor -DES, and one Digital Sun Sensor -DSS and backups) was no more considered safe because 3 of the six gyros on-board were out of order or very noisy. For that reason the MGM was implemented as default piloting mode. The MGM configuration was designed to pilot the ERS-2 using only one gyro plus the DES and the DSS modules. Scope of ZGM configuration was to extend the satellite lifetime by using the available gyros one at the time.



With the MGM, an increase of roughly 200 Hz was observed at the end of the qualification period. After the AOCS commissioning phase this parameter further evolved within the nominal range with a negligible impact on the data quality.

In MGM configuration, the gyro 5 was used until 7<sup>th</sup> October 2000 when it failed. From 10<sup>th</sup> October 2000 to 24<sup>th</sup> October 2000 the gyro 6 was used. This explains the decrease of roughly 100Hz in the CoG of the received spectrum. From 25<sup>th</sup> October 2000 to 17<sup>th</sup> January 2001 the gyro 1 was used to pilot the ERS-2 satellite. On 17<sup>th</sup> January 2001 the AOCS was upgraded. The new configuration allows piloting the satellite without gyroscopes. Unfortunately a failure of the Digital Earth Sensor (DES A-side) caused ERS-2 to enter in Safe-Mode on the same day. On 25<sup>th</sup> January 2001 gyro #1 also failed.

Satellite attitude was recovered on 5<sup>th</sup> February 2001 with a coarse attitude control mode (EBM). During the period of safe mode the spacecraft had drifted out of the nominal dead band by some 30 Km. The nominal orbit was reached on 6<sup>th</sup> February 2001. The EBM mode had a strong negative impact on the Scatterometer data quality and the dissemination of data products to end users was discontinued.

After that a series of AOCS upgrades has been implemented in order to improve the satellite attitude: on 30<sup>th</sup> March 2001 the Yaw steering law was re-introduced into the piloting function and on 7<sup>th</sup> June 2001 the Zero Gyro Mode (ZGM) has been implemented as nominal piloting mode. In ZGM the satellite attitude had an improvement in particular for the pitch and yaw error angle. This explains the reduction of the fluctuation in the received signal.

The CoG returns within its nominal value in February 2003 when the new ERS Scatterometer ground processor (ESACA) was put in operation (only for validation purposes) in Kiruna station. ESACA is able to compensate for errors in satellite attitude and to produce calibrated sigma noughts.

The evolution of the standard deviation of the CoG of the received spectrum was stable during the YSM phase. Small peaks are related with the events listed in Table 2. In MGM the evolution was within the nominal range while for the initial phase of the ZGM the performance was strong degraded. This because the on-ground Doppler filters was not able to compensate for the satellite degraded attitude. The introduction of the ESACA processor in February 2003 cured the problem.

**TABLE 1 ERS-2 Scatterometer AOCS depointing anomaly list**

Start of the anomaly			End of the anomaly			Remarks
24 <sup>th</sup> January	1996	9:10 a.m.	26 <sup>th</sup> January	1996	6:53 p.m.	AOCS depointing anomaly
14 <sup>th</sup> February	1997	1:25 a.m.	15 <sup>th</sup> February	1997	3:44 p.m.	AOCS depointing anomaly
3 <sup>rd</sup> June	1998	2:43 p.m.	6 <sup>th</sup> June	1998	12:47 a.m.	AOCS depointing anomaly
1 <sup>st</sup> September	1999	8:50 a.m.	2 <sup>nd</sup> September	1999	1:28 a.m.	
7 <sup>th</sup> October	2000	4:38 p.m.	10 <sup>th</sup> October	2000	4:49 p.m.	depointing anomaly gyro 5 failure
24 <sup>th</sup> October	2000	4:05 p.m.	25 <sup>th</sup> October	2000	12:05 p.m.	depointing anomaly gyro 6 failure
17 <sup>th</sup> January	2001		5 <sup>th</sup> February	2001		gyro 1 failure Satellite in safe mode

**TABLE 2 ERS-2 Scatterometer anomalies in the Doppler Compensation monitoring**

Date start	Year	Date stop	Year	Reason
26 <sup>th</sup> September	1996	27 <sup>th</sup> September	1996	Missing on-board Doppler coefficient (after cal. DC converter test period)
6 <sup>th</sup> June	1998	7 <sup>th</sup> June	1998	No Yaw Steering Mode (after depointing anomaly)
2 <sup>nd</sup> December	1998	3 <sup>rd</sup> December	1998	Missing on-board Doppler coefficients (after AMI anomaly number 228)
16 <sup>th</sup> February	2000	17 <sup>th</sup> February	2000	Fine Pointing Mode (FPM) (due to AOCS mono-gyro qualification period)
14 <sup>th</sup> April	2000	14 <sup>th</sup> April	2000	Fine Pointing Mode (FPM)
5 <sup>th</sup> July	2000	5 <sup>th</sup> July	2000	Fine Pointing Mode (FPM) after instrument switch-on
27 <sup>th</sup> September	2000	27 <sup>th</sup> September	2000	Fine Pointing Mode (FPM) to upload AOCS software patch
2 <sup>nd</sup> November	2000	2 <sup>nd</sup> November	2000	Fine Pointing Mode (FPM)
5 <sup>th</sup> December	2000	6 <sup>th</sup> December	2000	Fine Pointing Mode (FPM) due to orbital manoeuvre
6 <sup>th</sup> February	2001	30 <sup>th</sup> March	2001	Extra Backup Mode (EBM) coarse attitude control
30 <sup>th</sup> March	2001	17 <sup>th</sup> June	2001	ZGM-EBM coarse attitude control
17 <sup>th</sup> June	2001	21 <sup>st</sup> August	2003	ZGM phase. Error in yaw angle not corrected in the ground segment processor. Data shall be reprocessed with ESACA.
24 <sup>th</sup> March	2004	24 <sup>th</sup> March	2004	Fine Pointing Mode (FPM) due to orbital manoeuvre
25 <sup>th</sup> October	2004	27 <sup>th</sup> October	2004	Series of orbital manoeuvres (OCM and FPM)
10 <sup>th</sup> November	2004	11 <sup>th</sup> November	2004	Intense geomagnetic storm
8 <sup>th</sup> March	2005	8 <sup>th</sup> March	2005	orbital manoeuvre (OCM)
11 <sup>th</sup> March	2005	11 <sup>th</sup> March	2005	orbital manoeuvre (FPM)

During the cycles 105 - 109 the Doppler compensation evolution was very stable (see Figure 1). The CoG of the compensated signal was around 0 Hz for the Fore and Aft antenna and around 200 Hz for the Mid antenna. The standard deviation of the CoG was around 1500 Hz for the Fore and Aft antenna and around 2700 Hz for the Mid antenna. The small variation on the CoG evolution around 31 May is due to a slight degradation of the satellite pointing. Data gap in the statistics between 26 - 29 August 2005 is due to PCS System problem.

### ERS-2 WindScatterometer: DOPPLER COMPENSATION Evolution (UWI)

Least-square poly. fit fore beam      Center of gravity =  $-81.92 + (0.0349) \cdot \text{day}$       Standard Deviation =  $4665.3 + (-0.638) \cdot \text{day}$   
 Least-square poly. fit mid beam      Center of gravity =  $-774.7 + (0.3034) \cdot \text{day}$       Standard Deviation =  $5952.2 + (-0.774) \cdot \text{day}$   
 Least-square poly. fit aft beam      Center of gravity =  $-329.2 + (0.1188) \cdot \text{day}$       Standard Deviation =  $5419.1 + (-0.916) \cdot \text{day}$

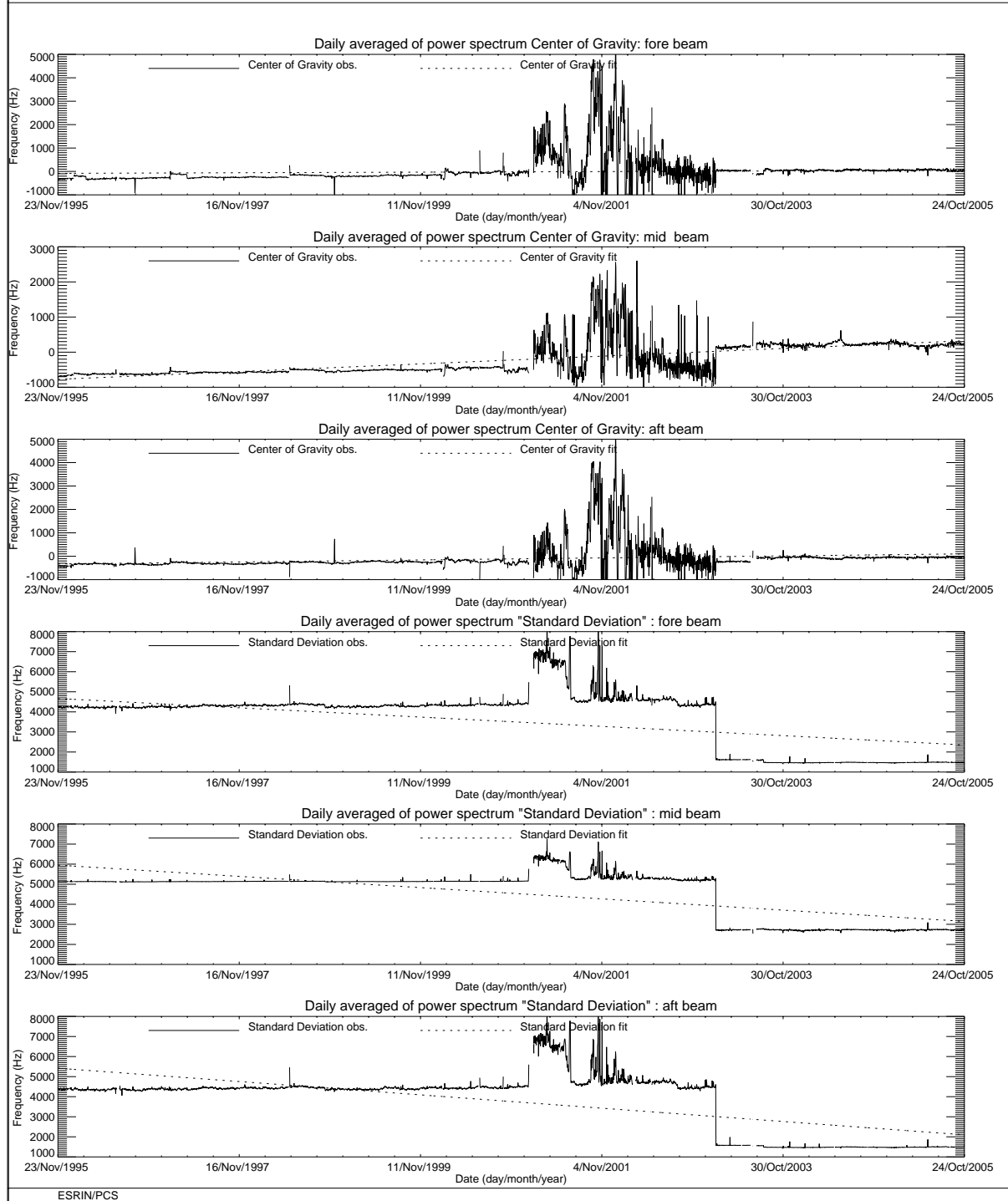
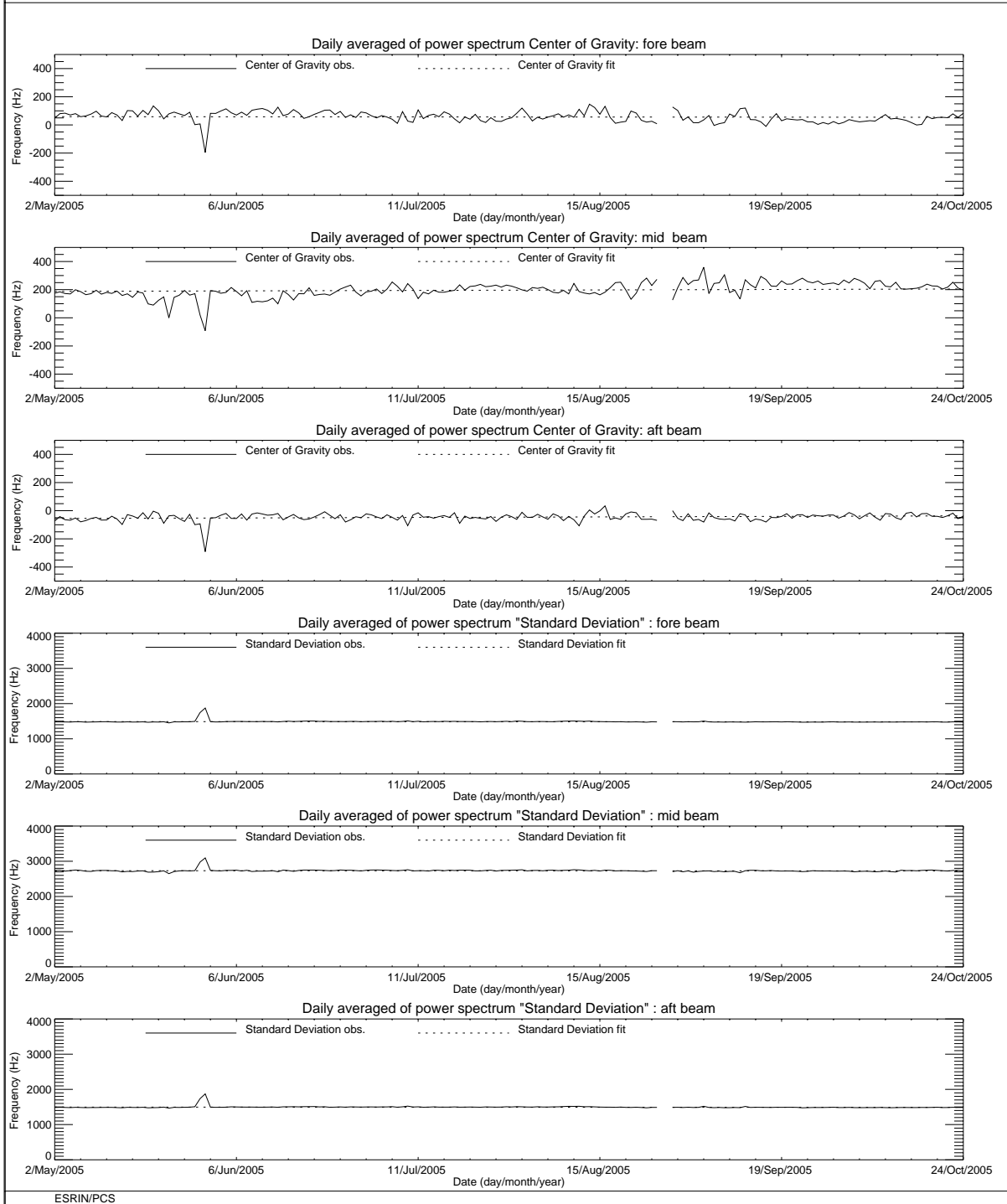


FIGURE 1 ERS-2 Scatterometer: Centre of Gravity and standard deviation of received power spectrum since the beginning of the mission.

### ERS-2 WindScatterometer: DOPPLER COMPENSATION Evolution (UWI)

Least-square poly. fit fore beam    Center of gravity =  $58.737 + (-0.025) \cdot \text{day}$     Standard Deviation =  $1488.7 + (-0.045) \cdot \text{day}$   
 Least-square poly. fit mid beam    Center of gravity =  $187.81 + (0.0968) \cdot \text{day}$     Standard Deviation =  $2732.9 + (-0.034) \cdot \text{day}$   
 Least-square poly. fit aft beam    Center of gravity =  $-55.24 + (0.1067) \cdot \text{day}$     Standard Deviation =  $1496.6 + (-0.039) \cdot \text{day}$



**FIGURE 2 ERS-2 Scatterometer: Centre of Gravity and standard deviation of received power spectrum during the cycles 105 - 109**

### 3.2 Noise power level I and Q channel

The results of the monitoring are shown in Figure 3 (long-term) and Figure 4 (cycles 105 – 109). The first set of three plots presents the noise power evolution for the I channel while the second set shows the Q channel. From the plots one can see that the noise level is more stable in the I channel than in the Q one. The I and Q receivers are inside the same box and any external interference should affect both channel. The fact that the receivers are closer to the ATSR-GOME electronics could have some impact but there is no clear explanation on that behavior. From 5<sup>th</sup> December 1997 until November 1998 some high peaks appear in the plots. These high values for the daily mean are due to the presence for these special days of a single UWI product with an unrealistic value in the noise power field of its Specific Product Header. The analysis of the raw data used to generate these products lead in all cases to the presence of one source packet with a corrupted value in the noise field stored into the source packet Secondary Header. The reason why noise field corruption is beginning from 5<sup>th</sup> December 1997 and last until November 1998 is at present unknown. It is interesting to note that at the beginning of December 1997, we started to get as well the corruption of the Satellite Binary Times (SBTs) stored in the EWIC product. The impact in the fast delivery products was the production of blank products starting from the corrupted EWIC until the end of the scheduled stop time. A change in the ground station processing in March 1998 overcame this problem.

Since 9<sup>th</sup> August 1998 until March 2000 some periods with a clear small instability in the noise power have been recognized, Table 3 gives the detailed list.

**TABLE 3 ERS-2 Periods with instability in the noise power**

Start date	Stop date	Year
9 <sup>th</sup> August	26 <sup>th</sup> October	1998
29 <sup>th</sup> November	6 <sup>th</sup> December	1998
23 <sup>rd</sup> December	24 <sup>th</sup> December	1998
7 <sup>th</sup> June	10 <sup>th</sup> June	1999
17 <sup>th</sup> August	22 <sup>nd</sup> August	1999
8 <sup>th</sup> September	9 <sup>th</sup> September	1999
3 <sup>rd</sup> October	8 <sup>th</sup> October	1999
16 <sup>th</sup> October	18 <sup>th</sup> October	1999
26 <sup>th</sup> October	28 <sup>th</sup> October	1999
25 <sup>th</sup> December	2 <sup>nd</sup> January	2000
10 <sup>th</sup> February	11 <sup>th</sup> February	2000
19 <sup>th</sup> March	26 <sup>th</sup> March	2000

To better understand the instability of the noise power the PCS has carried out investigations in the Scatterometer raw data (EWIC) to compute the noise power with more resolution. The result is that for the orbits affected by the instability the noise power had a decrease of roughly 0.7 dB for the fore and aft signals and a decrease of roughly 0.6 dB in the mid beam case (see the report for the cycle 42). The decrease of the noise power during the orbits affected by the instability is comparable with the decrease of the internal calibration level that occurred during the same orbits. The reason of this instability (linked to the AMI anomalies) is still unknown. On 28<sup>th</sup> February 2003 the Scatterometer receiver gain has been increased by 3 dB to optimize the usage of the on-board ADC converter. This explains the increase of the noise for the Fore and Aft beam channel. For the mid beam channel the noise still remains not measurable. The evolution of the noise power during the cycle 105 - 109 was stable (see Figure 3). The daily average for the Fore and Aft beam noise is around 1.7 ADC (I) and around 1.6 ADC (Q) respectively. For the Mid beam the noise is not measurable.

### ERS-2 WindScatterometer: NOISE Level Evolution (UWI)

Least-square line fit fore beam:  $I = 744.75 + (0.2391) \cdot \text{day}$

$Q = 694.62 + (0.2258) \cdot \text{day}$

I channel: No line fit standard deviation too high

Q channel: No line fit standard deviation too high

Least-square line fit aft beam:  $I = 743.94 + (0.2291) \cdot \text{day}$

Q channel: No line fit standard deviation too high

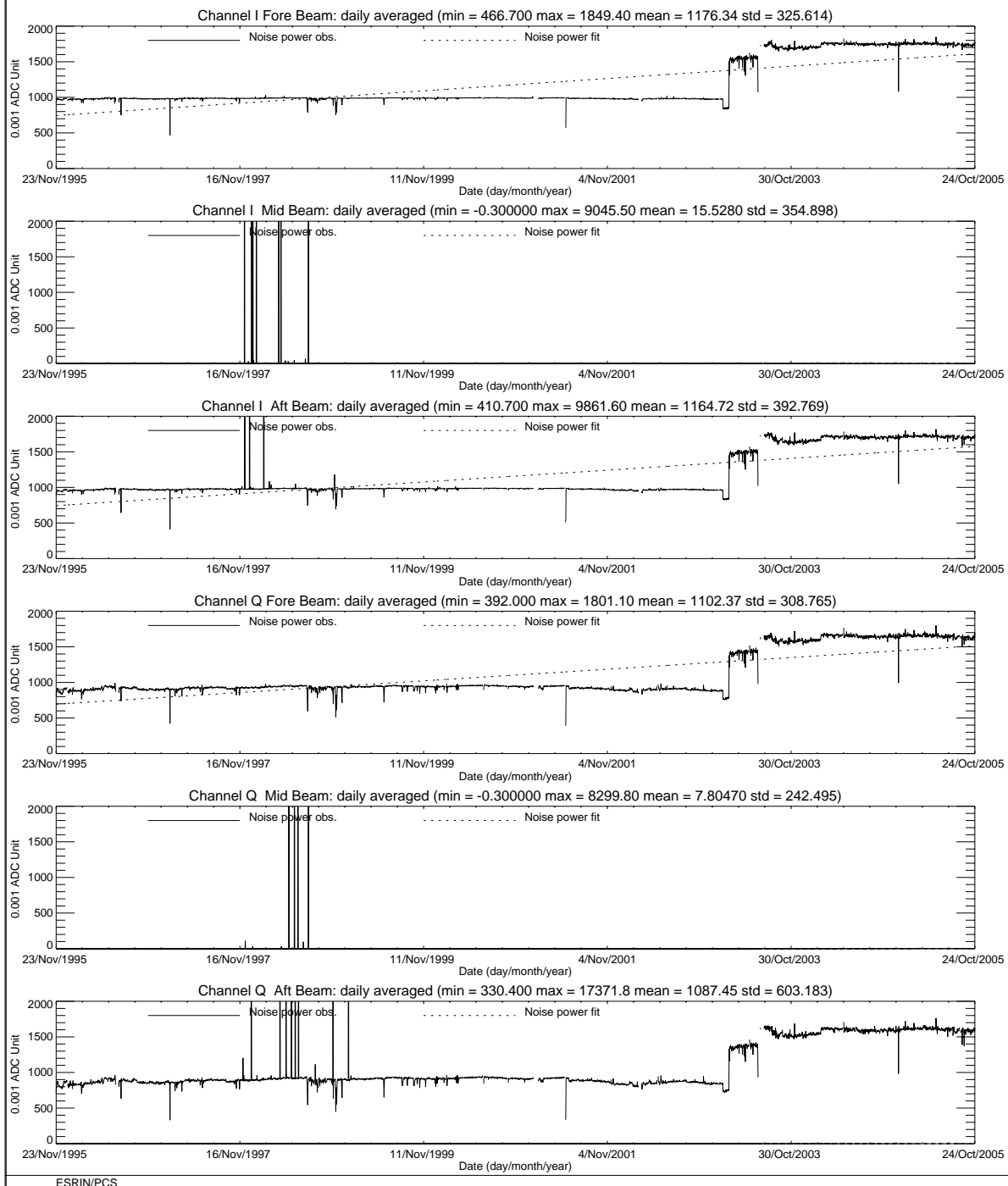


FIGURE 3 ERS-2 Scatterometer: noise power I and Q channel since the beginning of the mission.

### ERS-2 WindScatterometer: NOISE Level Evolution (UWI)

Least-square line fit fore beam:  $I = 1750.9 + (-0.074) \cdot \text{day}$   $Q = 1655.8 + (-0.156) \cdot \text{day}$   
 Least-square line fit mid beam:  $I = 0.0680 + (1.6296) \cdot \text{day}$   $Q = 0.0489 + (0.0002) \cdot \text{day}$   
 Least-square line fit aft beam:  $I = 1713.8 + (-0.069) \cdot \text{day}$   $Q = 1611.8 + (-0.167) \cdot \text{day}$

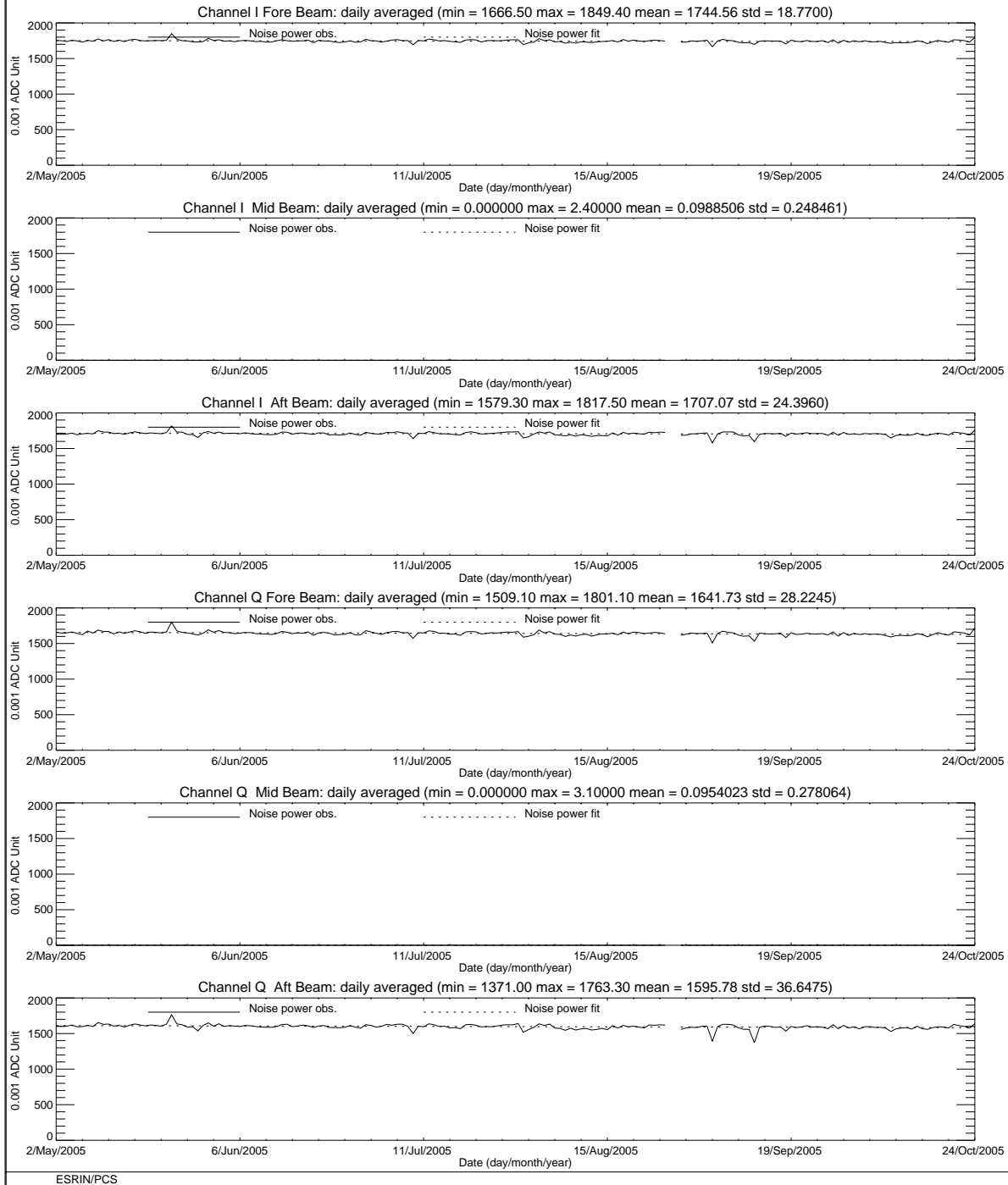


FIGURE 4 ERS-2 Scatterometer: noise power I and Q channel for cycles 105 - 109.



### 3.3 Power level of internal calibration pulse

For the internal calibration level, the results are shown in Figure 5 (long-term) and Figure 6 (cycles 105 – 109). The high value of the variance in the fore beam until August, 12<sup>th</sup> 1996 is due to the ground processing. In fact all the blank source packets ingested by the processor were recognized as Fore beam source packets with a default value for the internal calibration level. The default value was applicable for ERS-1 and therefore was not appropriate for ERS-2 data processing. On August 12<sup>th</sup>, 1996 a change in the ground processing LUT overcame the problem. Since the beginning of the mission a power decrease is detected. The power decrease is regular and affects the AMI when it is working in wind-only mode, wind/wave mode and image mode indifferently. The average power decrease is around 0.08 dB per cycle (0.0022 dB/day) and is clearer after August, 6<sup>th</sup> 1996 when the calibration subsystem has been changed. The reason of the power decrease is because the TWT is not working in saturation, so that a variation in the input signal is visible in the output. The variability of the input signal can be two-fold: the evolution of the pulse generator or the tendency of the switches between the pulse generator and the TWT to reset themselves into a nominal position. These switches were set into an intermediate position in order to put into operation the Scatterometer instrument (on 16<sup>th</sup> November 1995). To compensate for this decrease, on 26<sup>th</sup> October 1998 (cycle 37) 2.0 dB were added to the Scatterometer transmitted power and on 4<sup>th</sup> September 2002 (cycle 77) were added 3.0 dB. On 28<sup>th</sup> February 2003 (cycle 82) the Scatterometer receiver gain was increased by 3 dB to improve the usage of the on-board ADC converter. These events are clearly displayed by the large steps show in Figure 6.

Since 9<sup>th</sup> August 1998 until March 2000 the internal calibration level shows instability after an AMI or platform anomaly (see reports from cycle 35 to cycle 52). This instability is very well correlated with the fluctuations observed in the noise power. On 13<sup>th</sup> July 2000 a high peak (+3.5 dB) was detected in the transmitted power. This event has been investigated deeply by PCS and ESOC. The results of the analysis are reported in the technical note “ERS-2 Scatterometer: high peak in the calibration level” available in the PCS. The high transmitted power was detected after an arcing event which occurred inside the HPA. After that event the transmitted power had an average increase of roughly 0.14 dB.

During the cycle 105 – 109 the behavior of mean transmitted power evolution had for cycle 105 a decrease of 0.04 dB, for cycle 106 a decrease of 0.1 dB, for cycle 107 an increase of 0.1 db, for cycle 108 a decrease of 0.1 db and for cycle 109 a decrease of 0.1 dB. That value is closed to the nominal trend detected since August 1996 (about 0.1 dB per cycle). Only for cycle 107 we note an increase do not confirmed in the following cycles.



### ERS-2 WindScatterometer: Internal CALIBRATION Level Evolution (UWI)

Least-square polynomial fit fore beam	gain (dB) per day -0.0001	$1076.75 + (-0.0250833) \cdot \text{day}$
Least-square polynomial fit mid beam	gain (dB) per day -0.0001	$318.598 + (-0.00711484) \cdot \text{day}$
Least-square polynomial fit aft beam	gain (dB) per day -0.0001	$1064.34 + (-0.0236084) \cdot \text{day}$

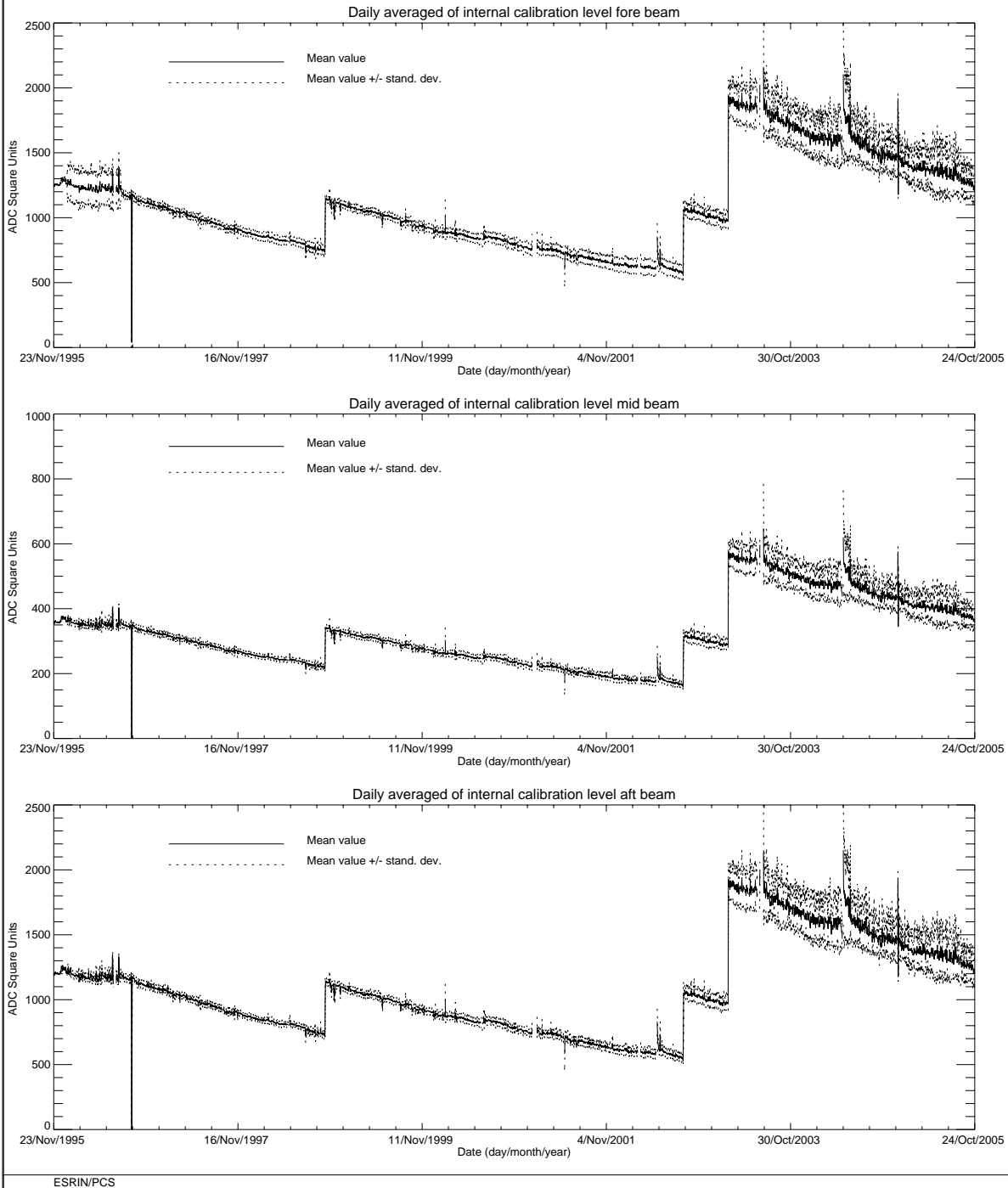


FIGURE 5 ERS-2 Scatterometer: power of internal calibration pulse since the beginning of the mission.

### ERS-2 WindScatterometer: Internal CALIBRATION Level Evolution (UWI)

Least-square polynomial fit fore beam	gain (dB) per day -0.0025	1390.32 +(-0.760813)*day
Least-square polynomial fit mid beam	gain (dB) per day -0.0024	410.407 +(-0.219432)*day
Least-square polynomial fit aft beam	gain (dB) per day -0.0025	1381.66 +(-0.759731)*day

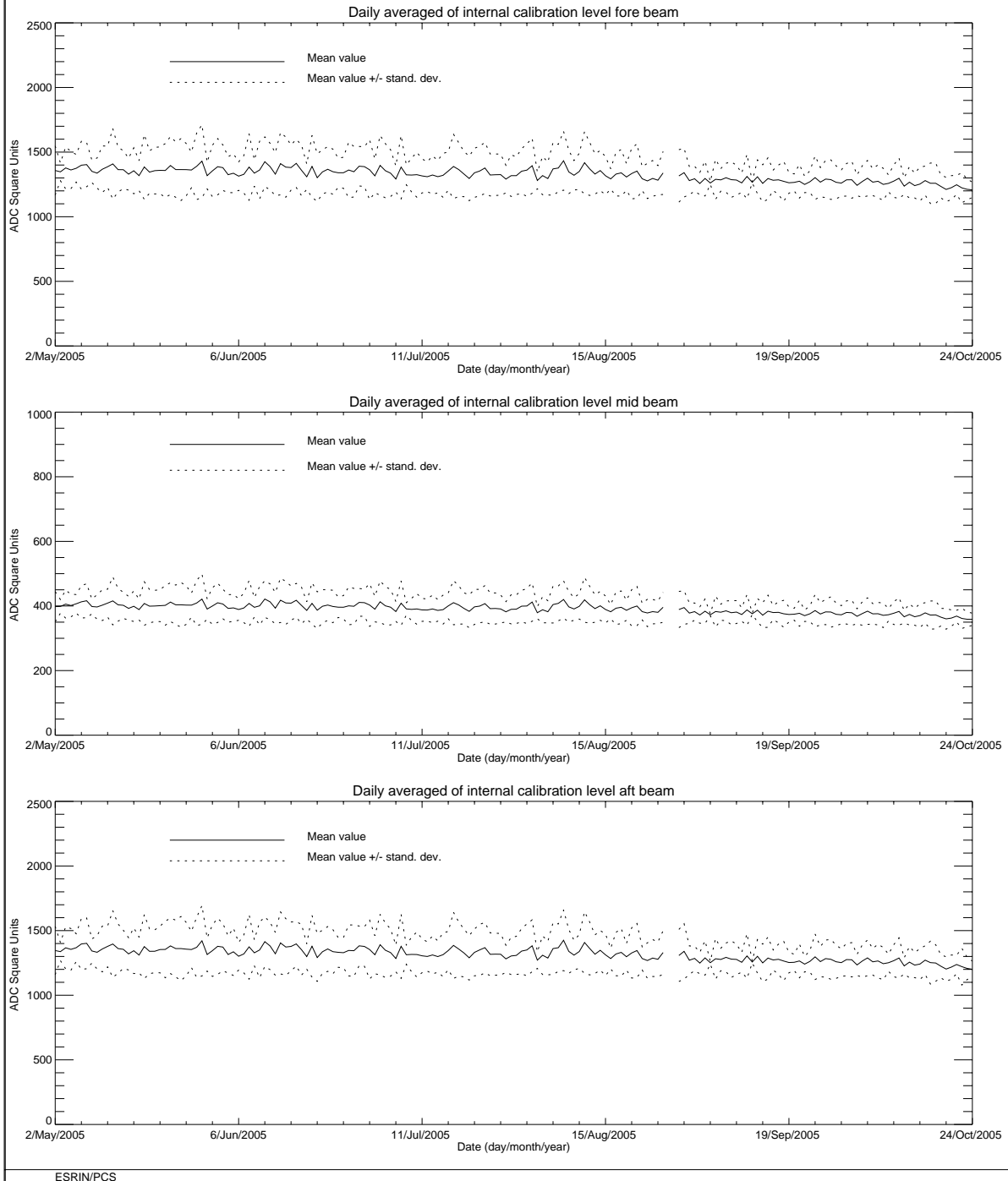


FIGURE 6 ERS-2 Scatterometer: power of internal calibration level cycles 105 - 109.

## **4 Products performance**

The PCS carries out a quality control of the winds generated from the WSCATT data. External contributions to this quality control (from ECMWF) but for all the specific information related to it can refer to cyclic ECMWF report published on web (<http://earth.esa.int/pcs/ers/scatt/reports/ecmwf>)

### **4.1 Products availability**

One of the most important points in the monitoring of the products performance is their availability. The Scatterometer is a part of ERS payload and it is combined with a Synthetic Aperture Radar (SAR) into a single Active Microwave Instrument (AMI). The SAR users requirements and the constraints imposed by the on-board hardware (e.g. amount of data that can be recorded in the on-board tape) set rules in the mission operation plan.

The principal rules that affected the Scatterometer instrument data coverage are:

- Over the Ocean the AMI is in wind/wave mode (Scatterometer with small SAR imaggettes acquired every 30 sec.) and the ATSR-2 is in low rate data mode.
- Over the Land the AMI is in wind only mode (only Scatterometer) and the ATSR-2 is in high rate mode. (Due to on board recorder capacity, ATSR-2 in high rate is not compatible with SAR wave imaggettes acquisitions.) This strategy preserves the Ocean mission.
- The SAR images are planned as consequence of users' request.

Moreover:

- since July 16<sup>th</sup> 2003 the ERS-2 Low Rate mission is continued within only the visibility of ESA ground stations over Europe, North Atlantic, the Arctic and western North America. The reason was the failure of both on-board tape recorders.
- During the cycles 64 – 92 (June 2001 since 25<sup>th</sup> February 2004) the AMI instrument was operated in wind/wave mode also over the land. The reason was because the SAR wave data was used to estimate the satellite mispointing along the full orbit. Since 25<sup>th</sup> February onwards the nominal mission scenario has been resumed, with the AMI instrument in wind only mode over the land (and consequently ATSR was operated again in High Rate over land). The mispointing performances (in particular the yaw error angle) along the full orbit are computing by analyzing the Scatterometer data.

In order to maximize the data coverage, after the on-board tape recorder failure, an upgrade of the ERS ground segment acquisition scenario has been performed.

In that framework the following has been implemented:

- Since September 7<sup>th</sup> 2003 the ground station in Maspalomas, Gatineau and Prince Albert are acquiring and processing data for all the ERS-2 satellite passes within the station visibility (apart from passes for which other satellites have an higher priority).
- To further increase the wind coverage of the North Atlantic area, since December 8<sup>th</sup>, 2003 is operative a new ground Station in West Freugh (UK) and data from this new station are available to the user since mid January 2004. Due to its location, the West Freugh acquisitions have some overlap with those from three other ESA stations, Kiruna, Gatineau or Maspalomas. The station overlap depends on the relative orbit of the satellite. Consequentially, overlapping wind Scatterometer LBR data may be included in two

products. Since the two products are generated at different ground stations the overlap may not be completely precise, with a displacement up to 12 Km and slight differences in the wind data itself.

- Since March, 3<sup>rd</sup> 2004, Matera station is acquiring low rate bit data for all the passes for which is planned a SAR acquisition. Some science data are produced and disseminated to users, Radar Altimeter data, Wave data and Scatterometer data are recorded on tapes and will be available off-line for re-processing. This means for the Scatterometer data coverage a very limited improvement due to the fact that are acquired only passes with some SAR activity.
- Since February 2005 a new acquisition station in Miami (US) is in operations. This new station allows a full data coverage of the Gulf of Mexico and part of the Pacific Ocean on the west Mexico coast.
- Since 25<sup>th</sup>, June 2005 a new acquisition stations have been put into operations in Beijing. It covers part of China and Oriental Asia.
- Since 5<sup>th</sup> July 2005 McMurdo ground station is operational in the South Pole. It covers all the Antarctic regions.

Figures 7, 8, 9, 10, 11 show the AMI operational modes for cycles respectively 105, 106, 107, 108, 109. Each segment of the orbit has different color depending on the instrument mode: brown for wind only mode, blue for wind-wave mode and green for image mode. The red and yellow colors correspond to gap modes (no data acquired). For all cycles the percentage of the ERS-2 AMI activity is shown in table 4, 5, 6, 7, 8. The values are within the nominal range.

**TABLE 4 ERS-2 AMI activity (cycle 105)**

Ami Mode	Ascending passes	Descending passes
<b>Wind and Wind-Wave</b>	<b>91.51 %</b>	<b>81.51%</b>
<b>Image</b>	<b>4.24 %</b>	<b>13.32 %</b>
<b>Gap and others</b>	<b>4.25 %</b>	<b>5.49 %</b>

**TABLE 5 ERS-2 AMI activity (cycle 106)**

Ami Mode	Ascending passes	Descending passes
<b>Wind and Wind-Wave</b>	<b>90.84 %</b>	<b>83.91%</b>
<b>Image</b>	<b>4.39 %</b>	<b>10.73 %</b>
<b>Gap and others</b>	<b>4.77 %</b>	<b>5.36 %</b>

**TABLE 6 ERS-2 AMI activity (cycle 107)**

Ami Mode	Ascending passes	Descending passes
<b>Wind and Wind-Wave</b>	<b>91.07 %</b>	<b>81.38%</b>
<b>Image</b>	<b>4.47 %</b>	<b>12.33 %</b>
<b>Gap and others</b>	<b>4.46 %</b>	<b>6.29 %</b>

**TABLE 7 ERS-2 AMI activity (cycle 108)**

Ami Mode	Ascending passes	Descending passes
<b>Wind and Wind-Wave</b>	<b>91.62 %</b>	<b>81.03%</b>
<b>Image</b>	<b>2.55 %</b>	<b>11.83 %</b>
<b>Gap and others</b>	<b>5.83 %</b>	<b>7.14 %</b>

**TABLE 8 ERS-2 AMI activity (cycle 109)**

Ami Mode	Ascending passes	Descending passes
<b>Wind and Wind-Wave</b>	<b>92.41 %</b>	<b>81.49%</b>
<b>Image</b>	<b>2.32 %</b>	<b>12.50 %</b>
<b>Gap and others</b>	<b>5.27 %</b>	<b>6.01 %</b>

Table 9 reports the major data lost due to the test periods, AMI and satellite anomalies or ground segment anomalies occurred after 6<sup>th</sup> August, 1996 (before that day for many times data were not acquired due to the DC converter failure).

**TABLE 9 ERS-2 Scatterometer mission major data lost (day or more) after 6<sup>th</sup>, August 1996**

Start date	Stop Date	Reason
September 23 <sup>rd</sup> , 1996	September 26 <sup>th</sup> , 1996	ERS 2 switched off due to a test period
February 14 <sup>th</sup> , 1997	February 15 <sup>th</sup> , 1997	ERS 2 switched off due to a depointing anomaly
June 3 <sup>rd</sup> , 1998	June 6 <sup>th</sup> , 1998	ERS 2 switched off due to a depointing anomaly
November 17 <sup>th</sup> , 1998	November 18 <sup>th</sup> , 1998	ERS 2 switched off to face out Leonide meteor storm
September 22 <sup>nd</sup> , 1999	September 23 <sup>rd</sup> , 1999	ERS 2 switched off due to Year 2000 certification test
November 17 <sup>th</sup> , 1999	November 18 <sup>th</sup> , 1999	ERS 2 switched off to face out Leonide meteor storm
December 31 <sup>st</sup> , 1999	January 2 <sup>nd</sup> , 2000	ERS 2 switched off Y2K transition operation
February 7 <sup>th</sup> , 2000	February 9 <sup>th</sup> , 2000	ERS 2 switched off due to new AOCS s/w up link
June 30 <sup>th</sup> , 2000	July 5 <sup>th</sup> , 2000	ERS 2 Payload switched off after RA anomaly
July 10 <sup>th</sup> , 2000	July 11 <sup>th</sup> , 2000	ERS 2 Payload reconfiguration
October 7 <sup>th</sup> , 2000	October 10 <sup>th</sup> , 2000	ERS 2 Payload switched off after AOCS anomaly
January 17 <sup>th</sup> , 2001	February 5 <sup>th</sup> , 2001	ERS 2 Payload switched off due to AOCS anomaly
May 22 <sup>nd</sup> , 2001	May 24 <sup>th</sup> , 2001	ERS 2 Payload switched off due to platform anomaly
May 25 <sup>th</sup> , 2001	May 25 <sup>th</sup> , 2001	AMI switched off due thermal analysis
November 17 <sup>th</sup> , 2001	November 18 <sup>th</sup> , 2001	ERS 2 switched off to face out Leonide meteor storm
November 27 <sup>th</sup> , 2001	November 28 <sup>th</sup> , 2001	ERS 2 payload off due to 1Gyro Coarse Mode commissioning
March 8 <sup>th</sup> , 2002	March 20 <sup>th</sup> , 2002	ERS 2 payload unavailability after RA anomaly
May 19 <sup>th</sup> , 2002	May 24 <sup>th</sup> , 2002	AMI switched off due to arc events
May 24 <sup>th</sup> , 2002	May 28 <sup>th</sup> , 2002	AMI partially switched off due to arc events
May 31 <sup>st</sup> , 2002	June 3 <sup>rd</sup> , 2002	Gatineau orbits partially acquired due to antenna problem
June 4 <sup>th</sup> , 2002	June 5 <sup>th</sup> , 2002	AMI partially switched-off due to arc events
July 25 <sup>th</sup> , 2002	July 25 <sup>th</sup> , 2002	AMI switched off HPA voltage too low
September 11 <sup>th</sup> , 2002	September 11 <sup>th</sup> , 2002	AMI switched off macrocommand transfer error
November 17 <sup>th</sup> , 2002	November 18 <sup>th</sup> , 2002	ERS-2 switched off to face out Leonide meteor storm
December 9 <sup>th</sup> , 2002	December 10 <sup>th</sup> , 2002	IDHT anomaly no data recorded on board
December 20 <sup>th</sup> , 2002	December 20 <sup>th</sup> , 2002	IDHT anomaly no data recorded on board
January 14 <sup>th</sup> , 2003	January 14 <sup>th</sup> , 2003	IDHT anomaly no data recorded on board
May 6 <sup>th</sup> , 2003	May 19 <sup>th</sup> , 2003	AMI off due to bus reconfiguration
June 22 <sup>nd</sup> , 2003	July 16 <sup>th</sup> , 2003	IDHT recorders test no data acquired
Since July 16 <sup>th</sup> , 2003		Regional Mission Scenario. Data available only within the visibility of ESA ground station
May 21 <sup>st</sup> , 2004	May 25 <sup>th</sup> , 2004	AMI in refuse mode due to excessive HPA arcing
June 22 <sup>nd</sup> , 2004	June 22 <sup>nd</sup> , 2004	AMI in refuse mode due to excessive HPA arcing
September 23 <sup>rd</sup> , 2004	September 24 <sup>th</sup> , 2004	AMI switched down
December 16 <sup>th</sup> , 2004	December 17 <sup>th</sup> , 2004	AMI memory test
December 26 <sup>th</sup> , 2004	December 26 <sup>th</sup> , 2004	IDHT anomaly. No data acquired
December 27 <sup>th</sup> , 2004	December 28 <sup>th</sup> , 2004	Payload off due to on board anomaly
January 23 <sup>rd</sup> , 2005	January 23 <sup>rd</sup> , 2005	AMI switched down (00.51 a.m. – 1.26 p.m.)
January 28 <sup>th</sup> , 2005	January 28 <sup>th</sup> , 2005	AMI switched down (01.29 a.m. – 10.39 a.m.)
February 26 <sup>th</sup> , 2005	February 26 <sup>th</sup> , 2005	AMI switched down (01.20 a.m. – 12.37 a.m.)
March 11 <sup>th</sup> , 2005	March 11 <sup>th</sup> , 2005	AMI switched down (02.43 p.m. – 05.41 p.m.)
Jun 20 <sup>th</sup> , 2005	Jun 21 <sup>st</sup> , 2005	AMI switched off caused by RBI status error (08:44 p.m. – 10:13 a.m.)

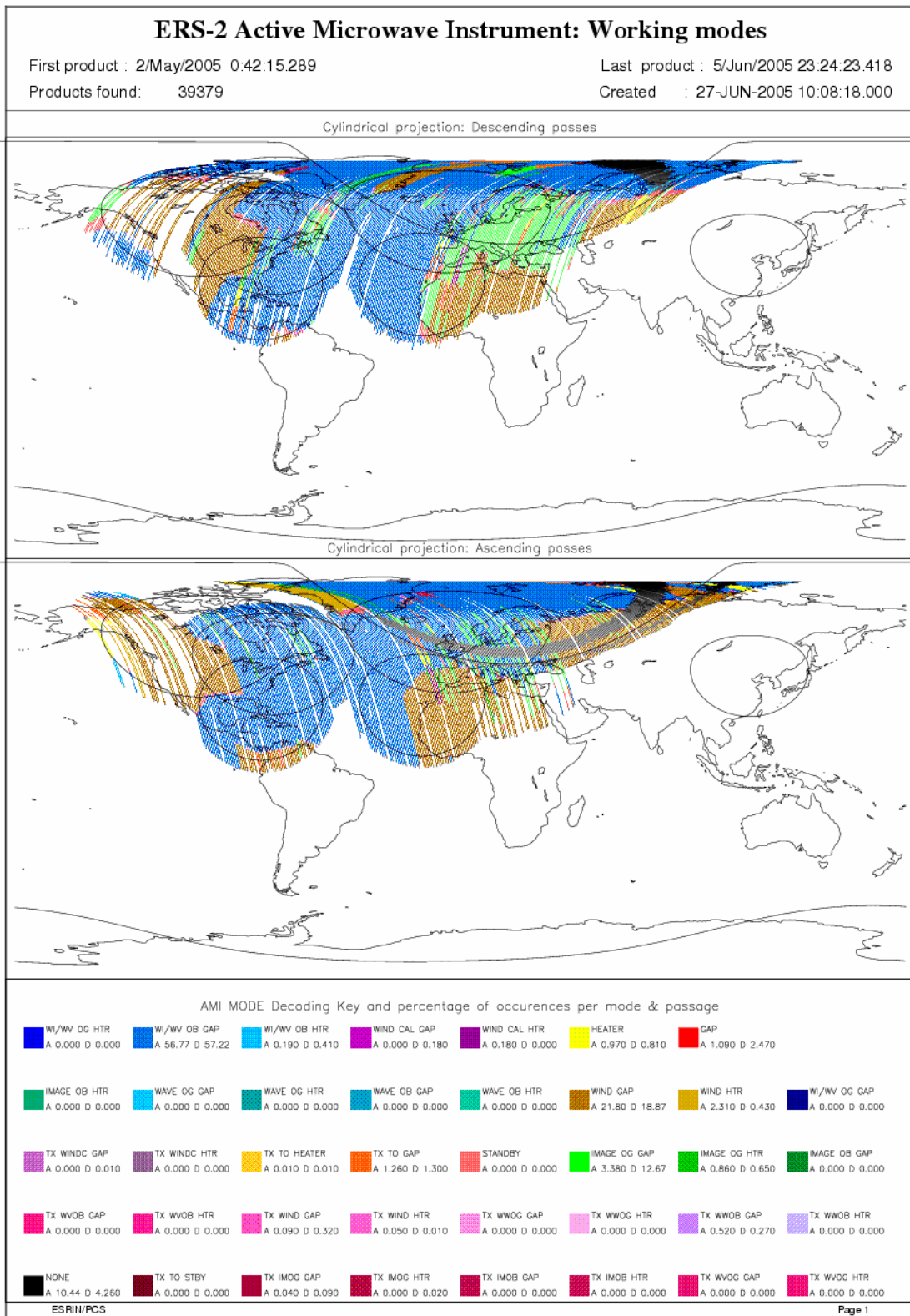


FIGURE 7 ERS-2 AMI activity during cycle 105.



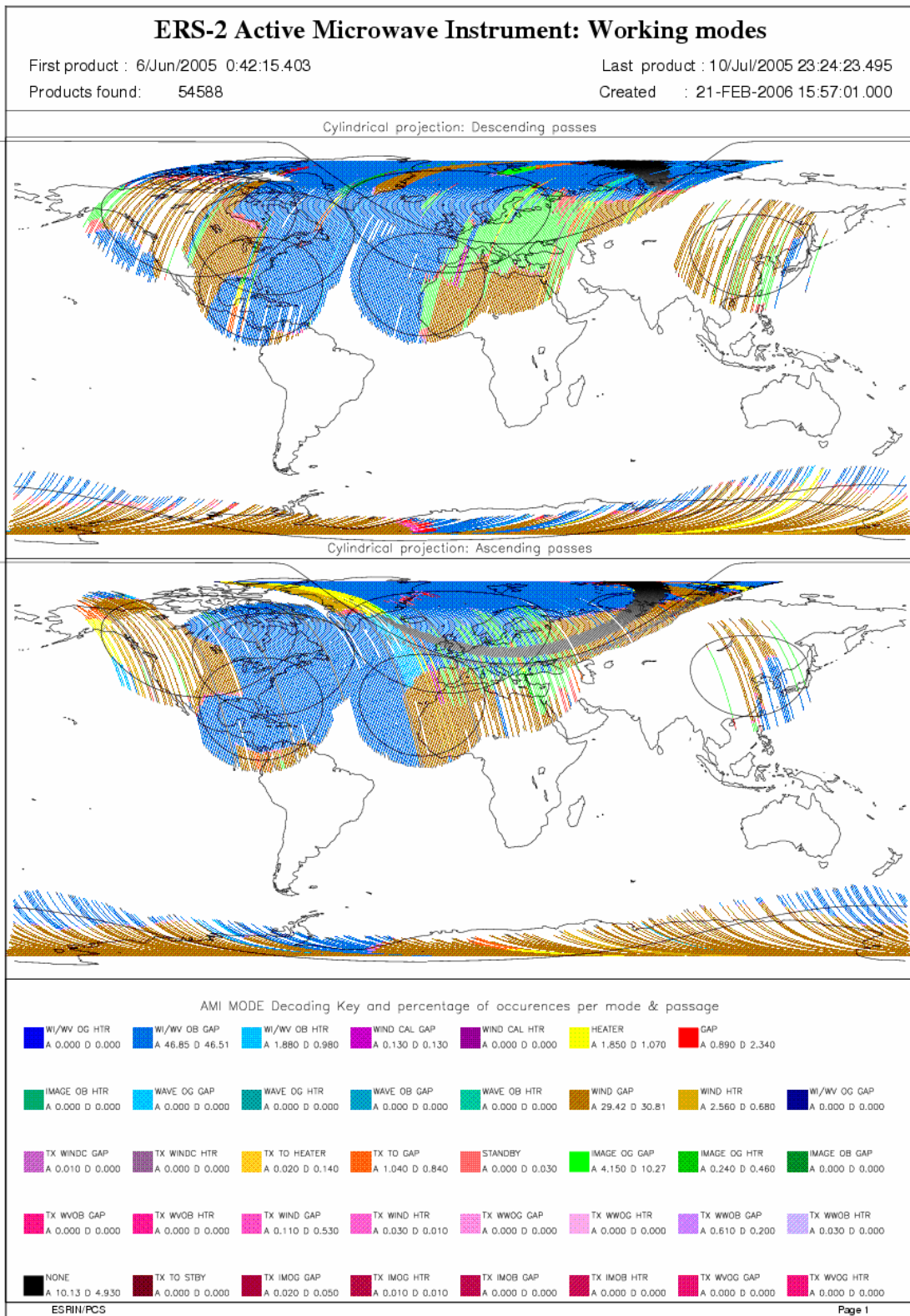


FIGURE 8 ERS-2 AMI activity during cycle 106.

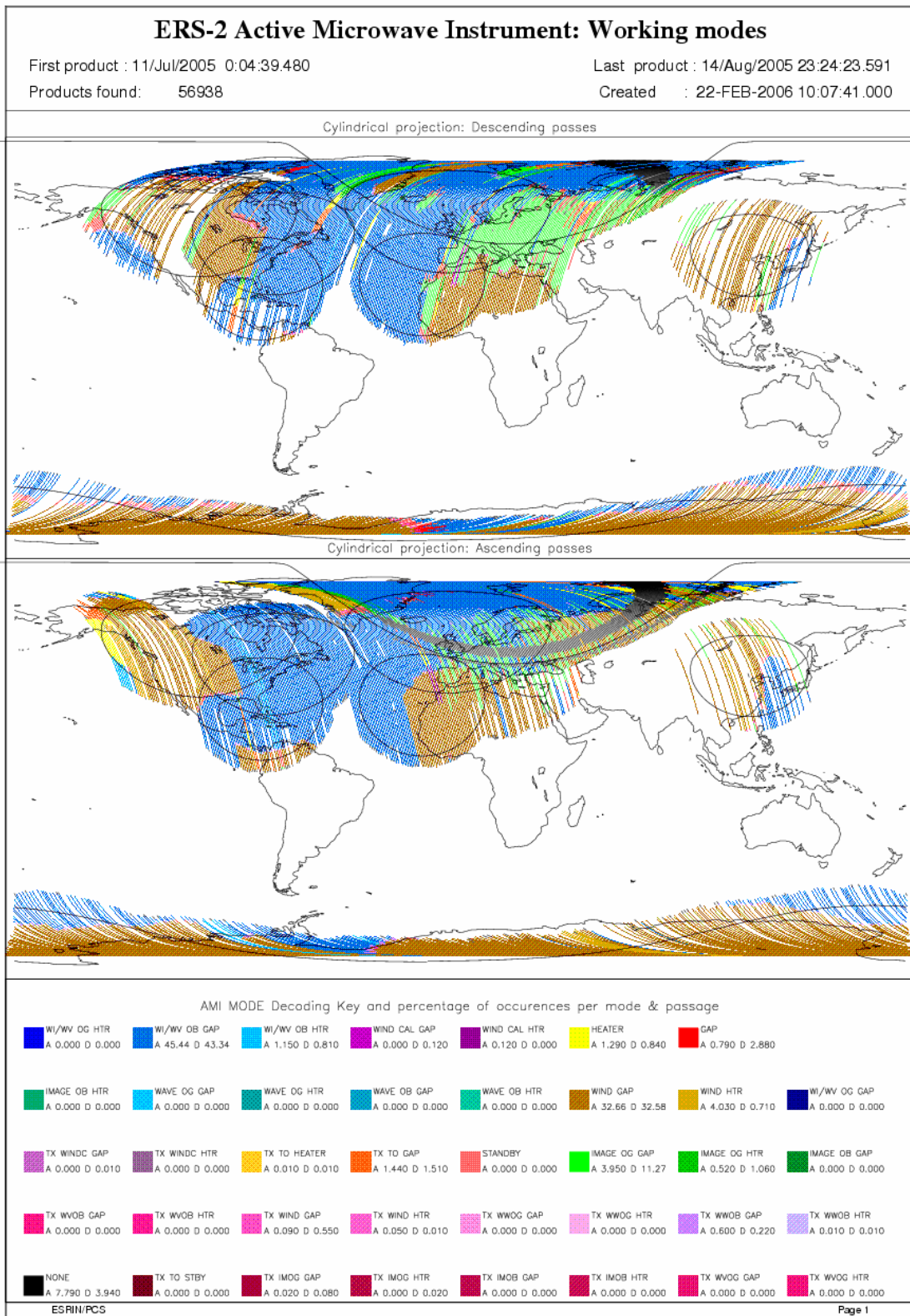


FIGURE 9 ERS-2 AMI activity during cycle 107.



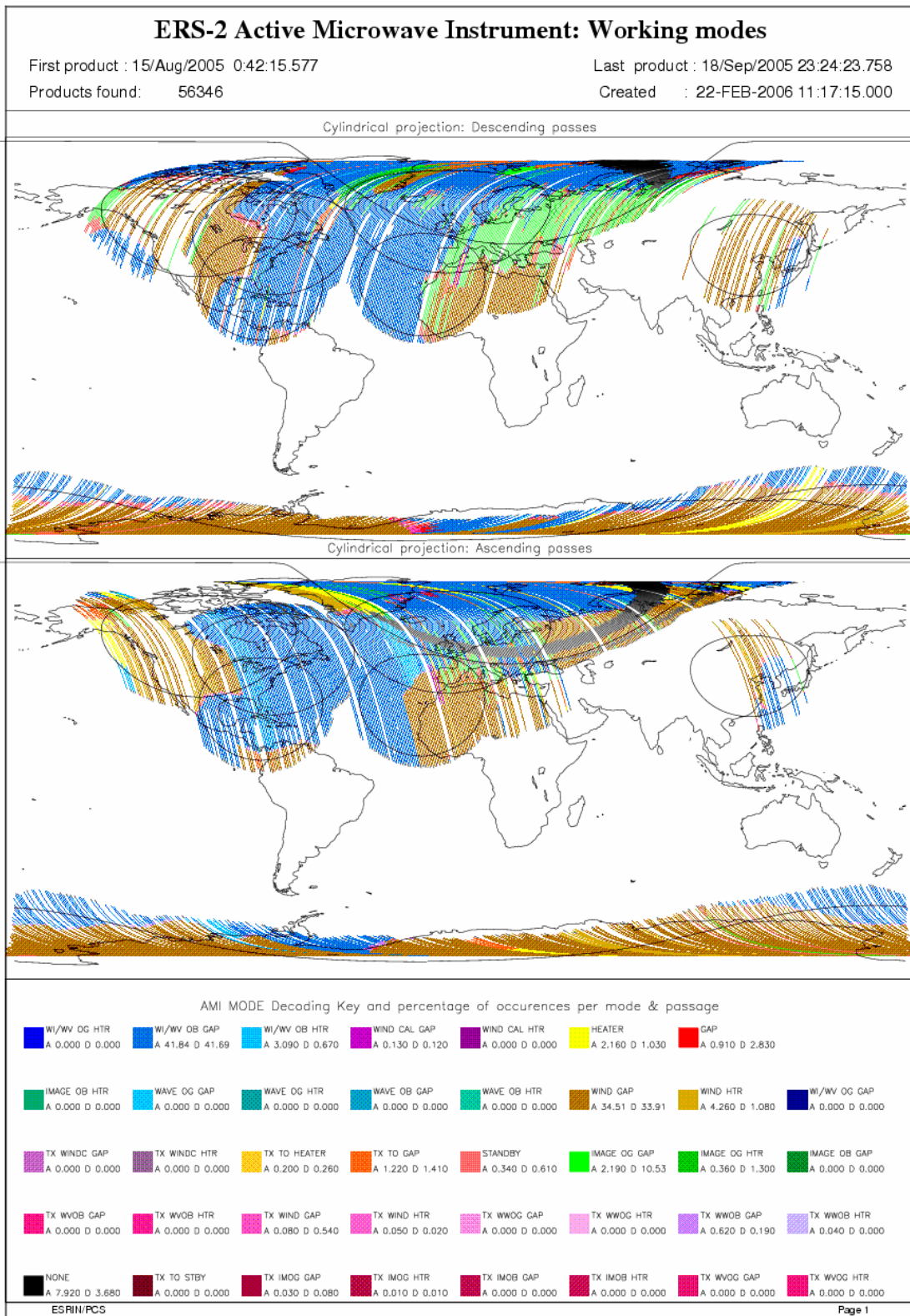


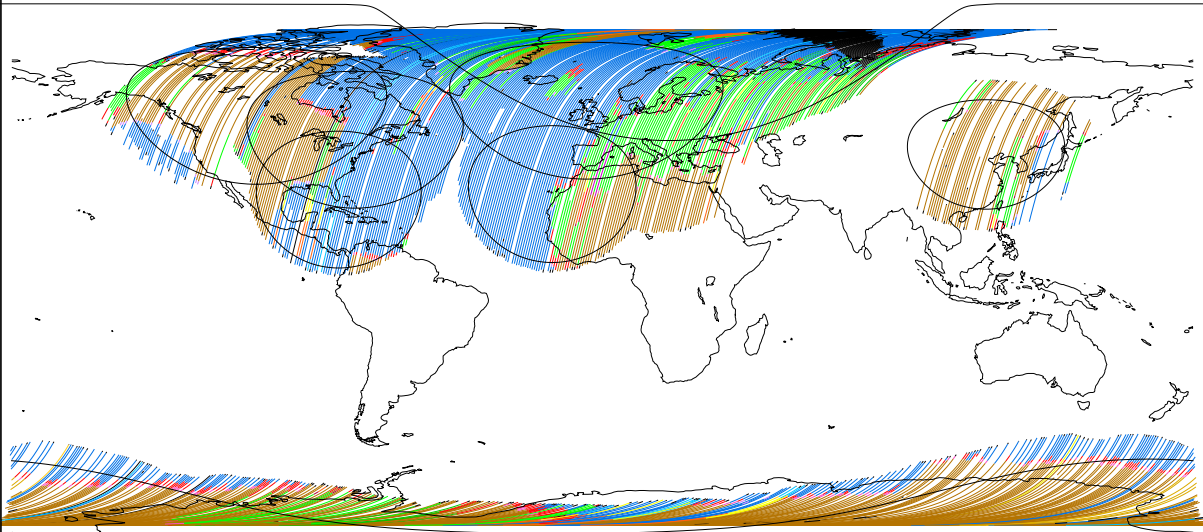
FIGURE 10 ERS-2 AMI activity during cycle 108.

### ERS-2 Active Microwave Instrument: Working modes

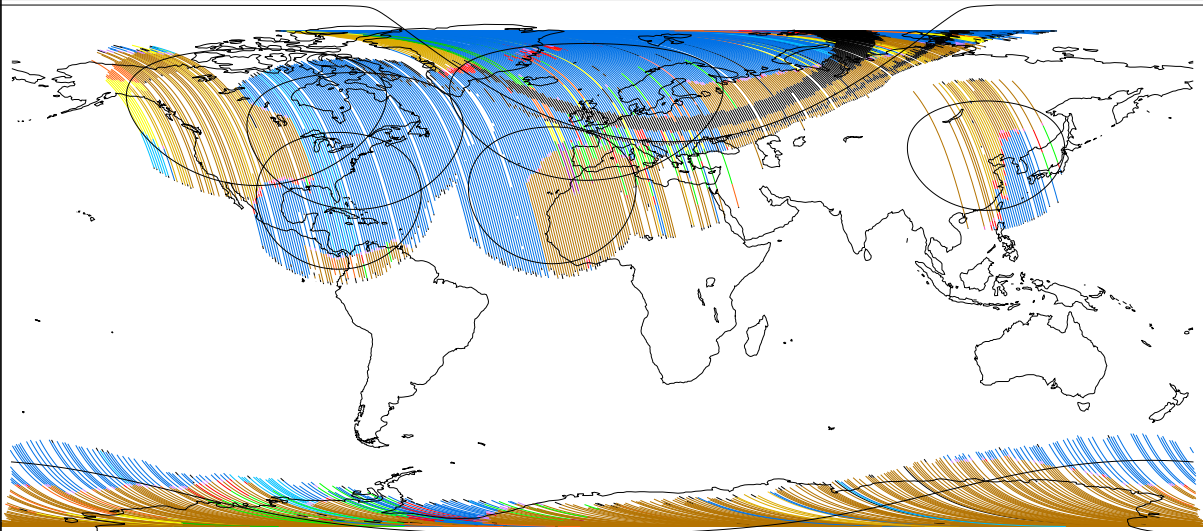
First product : 19/Sep/2005 0:04:39.744  
 Products found: 57466

Last product : 23/Oct/2005 23:24:07.982  
 Created : 22-FEB-2006 12:04:15.000

Cylindrical projection: Descending passes



Cylindrical projection: Ascending passes



#### AMI MODE Decoding Key and percentage of occurrences per mode & passage

WI/WV OG HTR A 0.000 D 0.000	WI/WV OB GAP A 44.12 D 42.69	WI/WV OB HTR A 2.100 D 1.210	WIND CAL GAP A 0.000 D 0.130	WIND CAL HTR A 0.120 D 0.000	HEATER A 1.510 D 0.690	GAP A 1.070 D 2.750	
IMAGE OB HTR A 0.000 D 0.000	WAVE OG GAP A 0.000 D 0.000	WAVE OG HTR A 0.000 D 0.000	WAVE OB GAP A 0.000 D 0.000	WAVE OB HTR A 0.000 D 0.000	WIND GAP A 32.86 D 32.73	WIND HTR A 5.300 D 0.460	WI/WV OG GAP A 0.000 D 0.000
TX WINDC GAP A 0.000 D 0.000	TX WINDC HTR A 0.010 D 0.000	TX TO HEATER A 0.220 D 0.160	TX TO GAP A 1.480 D 1.420	STANDBY A 0.030 D 0.000	IMAGE OG GAP A 1.680 D 10.92	IMAGE OG HTR A 0.640 D 1.580	IMAGE OB GAP A 0.000 D 0.000
TX WVOB GAP A 0.000 D 0.000	TX WVOB HTR A 0.000 D 0.000	TX WIND GAP A 0.100 D 0.550	TX WIND HTR A 0.040 D 0.010	TX WVOG GAP A 0.000 D 0.000	TX WVOG HTR A 0.000 D 0.000	TX WVOB GAP A 0.590 D 0.190	TX WVOB HTR A 0.020 D 0.000
NONE A 8.030 D 4.400	TX TO STBY A 0.000 D 0.000	TX IMOG GAP A 0.020 D 0.090	TX IMOG HTR A 0.000 D 0.010	TX IMOB GAP A 0.000 D 0.000	TX IMOB HTR A 0.000 D 0.000	TX WVOG GAP A 0.000 D 0.000	TX WVOG HTR A 0.000 D 0.000

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FIGURE 11 ERS-2 AMI activity during cycle 109.

## 4.2 PCS Geophysical Monitoring

The routine analysis is summarized in the plots of figure 12; from top to bottom:

- the monitoring of the valid sigma-nought triplets per day.
- the evolution of the wind direction quality. The ERS wind direction (for all nodes and only for those nodes where the ambiguity removal has worked properly) is compared with the ECMWF forecast. The plot shows the percentage of nodes for which the difference falls in the range -90.0, +90.0 degrees.
- the monitoring of the percentage of nodes whose ambiguity removal works successfully.
- the comparison of the wind speed deviation: (bias and standard deviation) with the ECMWF forecast.

The results since August 6<sup>th</sup>, 1996 until the beginning of the operation with the Zero Gyro Mode (ZGM) in January 2001 can be summarized as:

- High quality wind products has been distributed since Mid March 1996 (end of calibration and validation phase)
  - The number of valid sigma-nought distributed per day was almost stable with a small increase after June 29<sup>th</sup>, 1999 due to the dissemination in fast delivery of the data acquired in the Prince Albert station (Canada).
  - The wind direction is very accurate for roughly 93% of the nodes, the ambiguity removal processing successfully worked for more than 90.0% of the nodes.
  - The UWI wind speed shows an absolute bias of roughly 0.5 m/s and a standard deviation that ranges from 2.5 m/s to 3.5 m/s with respect to the ECMWF forecast.
  - The wind speed bias and its standard deviation have a seasonal pattern due to the different winds distribution between the winter and summer season.
  - Two important changes affect the speed bias plot.
  - the first is on June 3<sup>rd</sup>, 1996 due to the switch from ERS-1 to ERS-2 data assimilation in the meteorological model.
  - the second which occurred at the beginning of September 1997, is due to the new monitoring and assimilation scheme in ECMWF algorithms (4D-Var).
  - Since 19<sup>th</sup> April 1999 two set of meteo-table (meteorological forecast centred at 00:00 and 12:00 of each day) are used in the ground processing. This allowed the processing of wind data with 18 and 24 hours meteorological forecast instead of the 18, 24, 30 36 hours forecast. The comparison between data processed with the 18-24 hours forecast instead of 30-36 hours forecast shown an increase in the number of ambiguity removed nodes with a neutral impact in the daily statistics.
  - The mono-gyro AOCS configuration (see report for cycle 50) that was operative from 7<sup>th</sup> February 2000 to 17<sup>th</sup> January 2001 did not affect the wind data performance.
- During the Zero Gyro Mode (ZGM) phase the dissemination of the fast delivery Scatterometer data to the users has been interrupted on 17<sup>th</sup> January 2001 due to degraded quality in sigma noughts and winds. The satellite attitude in ZGM is slightly degraded and the “old” ground processor was not able to produce calibrated data anymore. For that reason a re-design of the entire ground processing has been carried out and since August 21<sup>st</sup> 2003 the new processor named ERS Scatterometer Attitude Corrected Algorithm (ESACA) is operative in all the ESA ground station and data was redistributed to the user.
- Although for a long period data was not distributed, the PCS has monitored the data quality (as shown in Figure 9) and the results during that period can be summarized as:

At the beginning of the ZGM (January 2001 - end July 2001) the number of valid nodes has clear drop from 190000 per day to 9000 per day. This because the satellite attitude was strong degraded and the received signal had a very high Kp figure (in particular for the far range nodes). For the valid nodes, due to no calibrated sigma nought, the quality of the wind was very poor, the distance from the cone was high and the wind speed bias was above 1.5 m/s.

At the end of July 2001 the ZGM has been tuned and the satellite attitude had an improvement. This explains the increase of the number of valid nodes (returned around the nominal level) and the improvements in the wind speed bias (around 0.5 m/s).

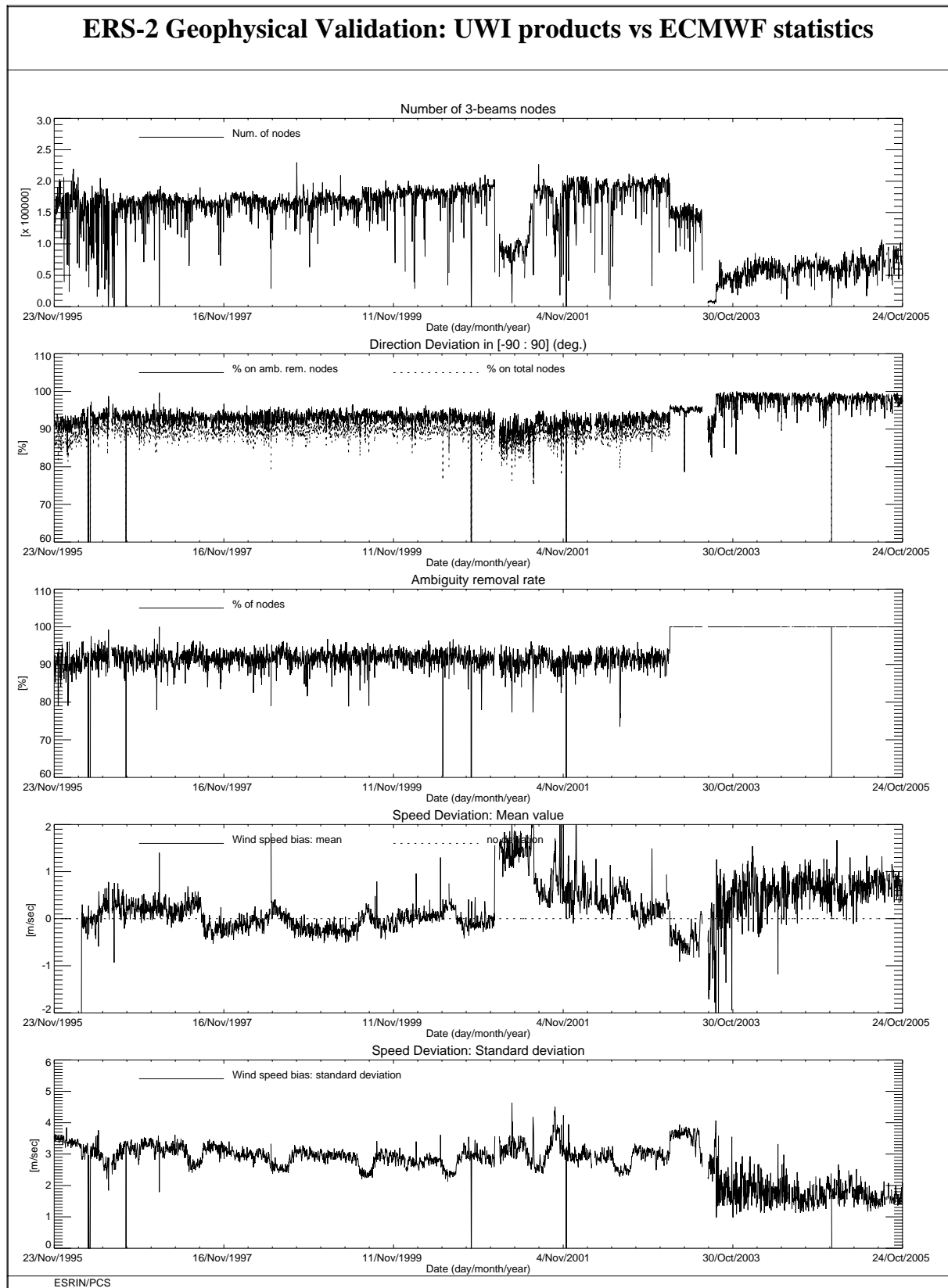
On 4<sup>th</sup> February 2003, a beta version of the new ESACA processor has been put in operation in Kiruna for validation and the monitoring of the data quality has been done only for the new ESACA data. The number of valid nodes slight decreased because Kiruna station process only 9 of 14 orbits per day. The wind speed direction deviation had a clear improvement because ESACA implements a new ambiguity removal algorithm (MSC) and the ambiguity removal rate is now stable at 100% (the MSC is able to remove ambiguity for all the nodes). The wind speed bias had a clear drop from 0.5 to -0.5 m/s. That value is closer to the nominal one (around -0.2 m/s). As reported in the previous cyclic reports the beta version of ESACA had some calibration problem for the near range nodes and this explains why the data quality does not match exactly the one obtained in the nominal YSM. That problem has been overcome with the final release of the ESACA processor put into operation on August 21<sup>st</sup> 2003. On June 22<sup>nd</sup> the failure of the on-board tape recorder discontinued the ERS global mission (see section 4.1) and this explains the low number of valid nodes available after that day.

The performances of ESACA winds delivered between August 2003 and September 2004 are affected by land contamination. Around costal zones many Sea nodes have a strong contribution of Land backscattering and the retrieved wind is not correct. An optimization of the Land/Sea flag in the ground processing has been carried out during the cycle 98. In the statistics computed by PCS on the fast delivered winds the Land contamination has been removed by using a refined Land/Sea mask. Also the ice contamination has been removed with a simple geographical filter. With these new setting the PCS statistics are very similar to the ones reported by ECMWF.

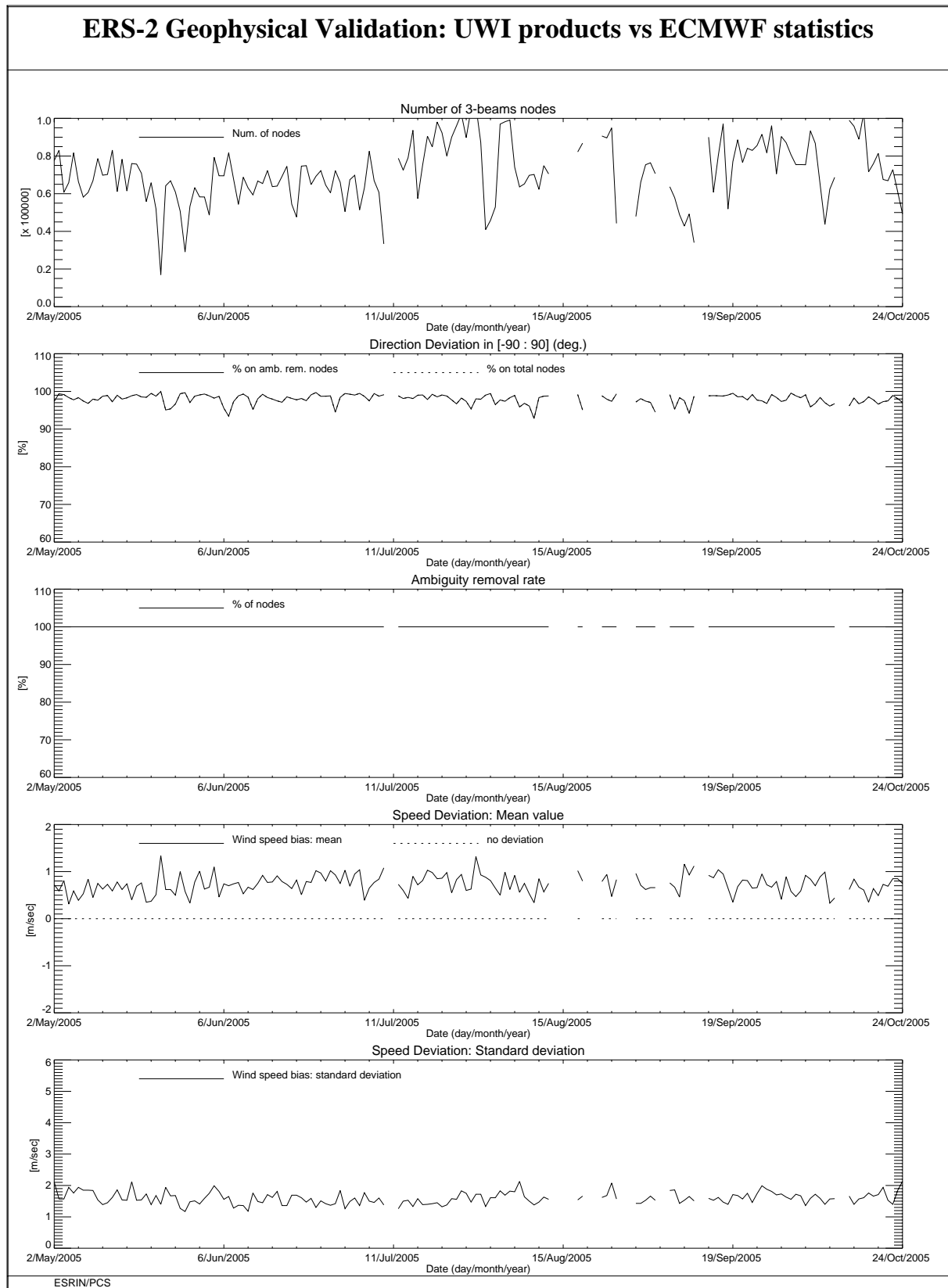
For cycles 105 – 109 the wind performance was very stable except between 15<sup>th</sup> – 18<sup>th</sup> August 2005 due to ESRIN network problem; for the period 22<sup>nd</sup> – 23<sup>rd</sup> August 2005 and 28<sup>th</sup> – 29<sup>th</sup> August 2005 due to PCS System problem. This caused a degradation of the wind field quality with a temporary de-aliasing problem of UWI products. For the rest of period the wind speed bias (UWI vs 18 or 24 hour forecast) was closed in nominal trend and the speed bias standard deviation was in nominal seasonal trend.

The wind direction deviation was good. The performance for UWI wind direction follows the seasonal trend. The standard deviation of UWI wind speed has increased, the reason could be related to a less mild wind climate.





**FIGURE 12 ERS-2 Scatterometer: wind products performance since the beginning of the mission.**



**FIGURE 13 ERS-2 Scatterometer: wind products performance for cycles 105 - 109.**

### **4.3 ECMWF Geophysical Monitoring**

The quality of the UWI product was monitored at ECMWF for cycle 105, 106, 107, 108 and 109. Results were compared to those obtained from the previous cycle, as well for data received during the nominal period in 2000 (up to cycle 59). No corrections for duplicate observations were applied.

The detailed information related to these cycles are reported in ECMWF reports published on web site (<http://earth.esa.int/pcs/ers/scatt/reports/ecmwf>)

#### **4.3.1 Distance to cone history**

Refer to ECMWF report published on web (<http://earth.esa.int/pcs/ers/scatt/reports/ecmwf>)

#### **4.3.2 UWI minus First-Guess history**

Refer to ECMWF report published on web (<http://earth.esa.int/pcs/ers/scatt/reports/ecmwf>)

#### **4.3.3 Scatter plots**

Refer to ECMWF report published on web (<http://earth.esa.int/pcs/ers/scatt/reports/ecmwf>)

## 5 Yaw error angle estimation

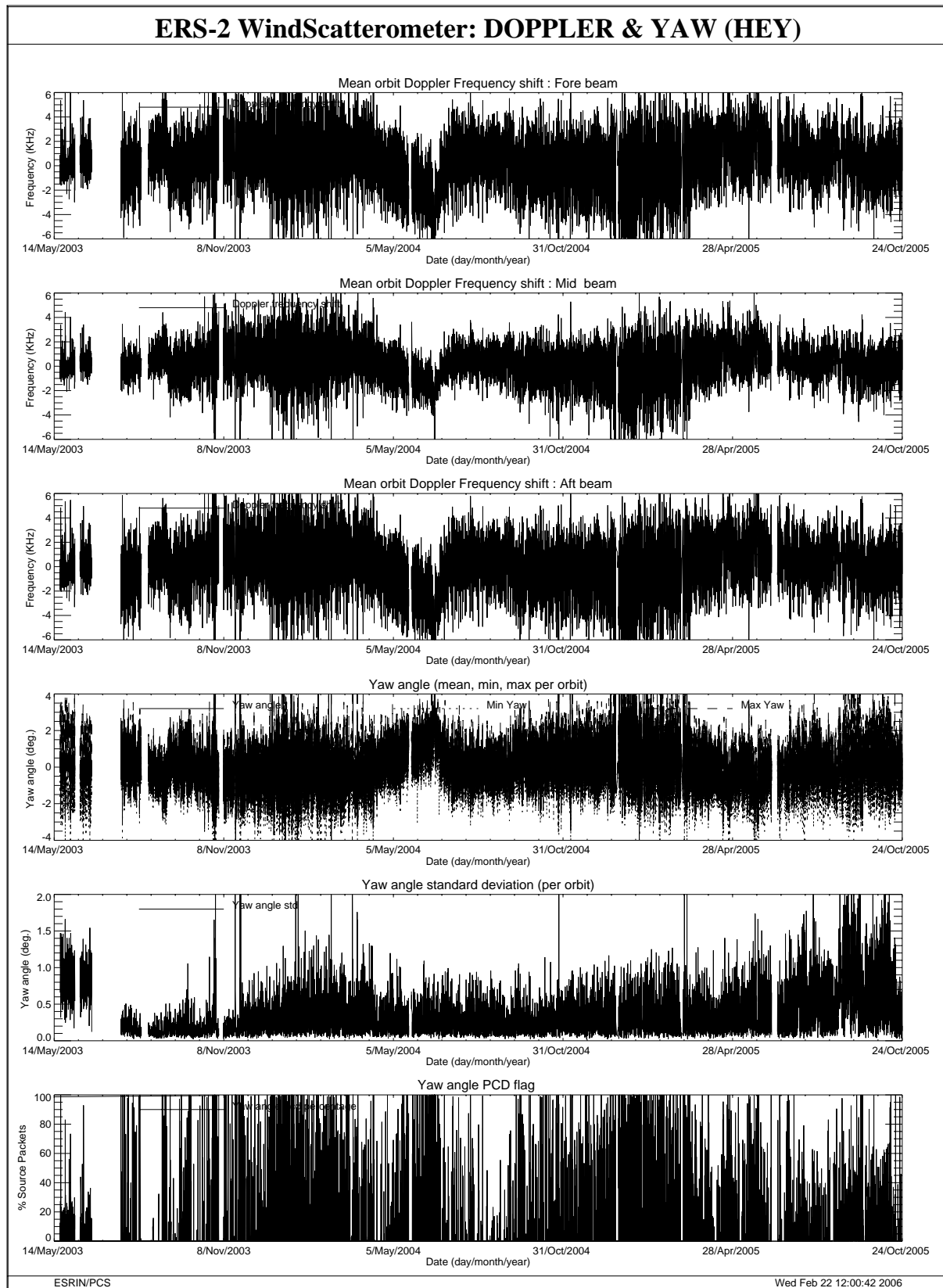
The yaw error angle estimation is computed on-ground by the ESACA processors. The full set of results of the yaw processing is stored in an internal ESA product named HEY (Helpful ESA Yaw) disseminated from the ground station to ESRIN. The estimation of the yaw error angle is based on the Doppler shift measured on the received echo. That estimation can be done with a good accuracy only for small yaw error angle (in the range between +/-4 deg.). Above that range, due to high Doppler frequency shift the signal spectrum is outside the receiver bandwidth and the yaw estimation is strongly degraded. Details regarding the yaw processing can be found on the following document (chapter 9): <http://earth.esa.int/pcs/ers/scatt/articles/soamain-030521.pdf>.

The yaw error angle estimation aims to compute the correct acquisition geometry for the three Scatterometer antenna throughout the entire orbit. The Yaw error angle information is used in the radar equation to derive the calibrated backscattering (sigma nought) from the Earth surface and to select the echo samples associated to one node. In ESACA the definition of the node position is as the one adopted in the old processor (for details see: [http://earth.esa.int/pcs/ers/scatt/articles/scatt\\_work98\\_processing.pdf](http://earth.esa.int/pcs/ers/scatt/articles/scatt_work98_processing.pdf)). In such way the distance between the nodes (both along and across track) is kept constant (25 Km) and what is changing in function of the yaw error angle is the number of echo samples that contributes to the node calculation and the incidence angle of the measurement. This because the three Scatterometer antennae could see the node with a different geometry due to an arbitrary variation of the yaw angle along track. The number of samples that actually contributes to a node and the yaw flag can be retrieved from the UWI Data Set Record (DSR) product. For that reason the definition of few fields in the UWI product has been updated. For details see the Scatterometer cyclic report - cycle 90 -. The Figure 26 (since beginning of HEY dissemination) and Figure 27 (cycle) show for each orbit the average Doppler frequency shift (first 3 plots Fore Mid and Aft antenna), the minimum, maximum and mean yaw (fourth plot), the yaw standard deviation (fifth plot) and the percentage of source packets acquired with a yaw error angle outside the range +/- 2 degrees (sixth plot).

On average the yaw evolution is within the specification for the ESACA processor to assure calibrated data. The evolving yaw bias occurred in June 2004 has been reported to the flight segment and corrective actions have been put in place to compensate for.

The result of the yaw monitoring for cycles 105, 106, 107, 108, 109 is a yaw error angle within the expected nominal range (+/- 2 degrees) with an average level around 0 deg. for most of the orbits. The gap shown in the plot is related to an error in the generation of the statistic; the data dissemination was nominal. An increase of degradation in the yaw performance was noted starting from cycle 107.





**FIGURE 14** Doppler frequency shift and Yaw error angle evolution since August 2003.

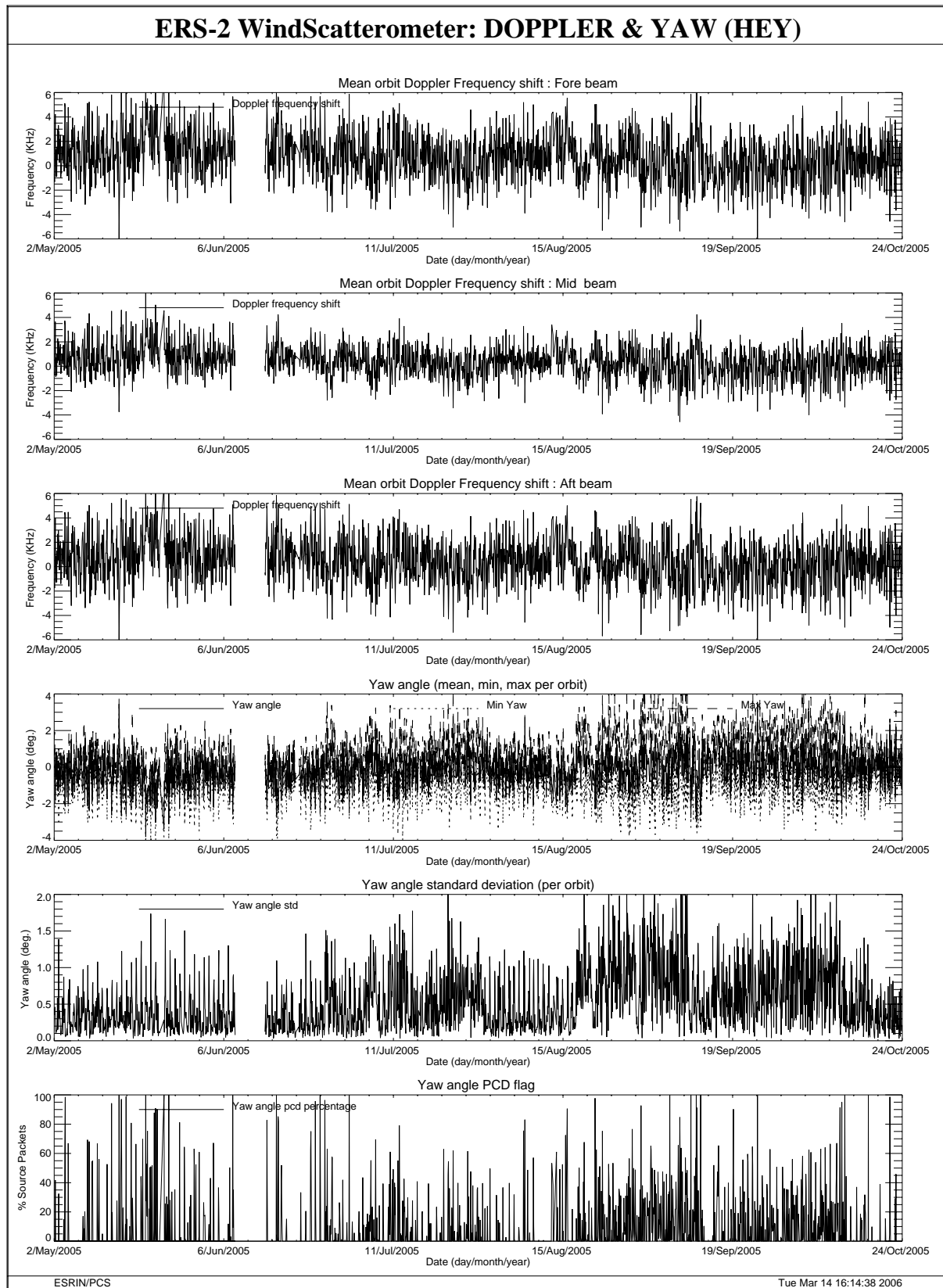


FIGURE 15 Doppler frequency shift and Yaw error angle evolution cycles 105 - 109.